Precise Orbit Determination

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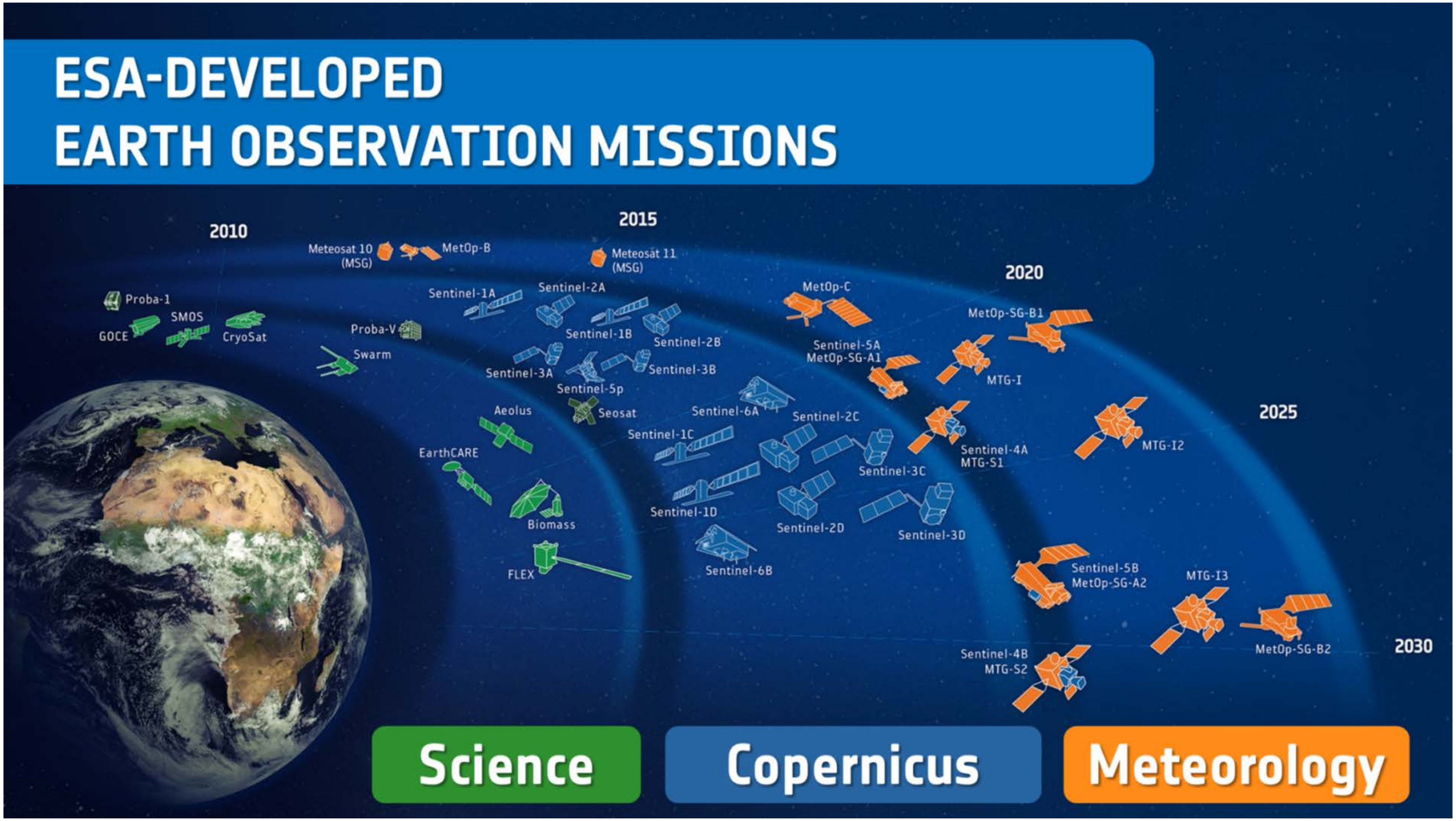
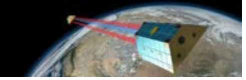
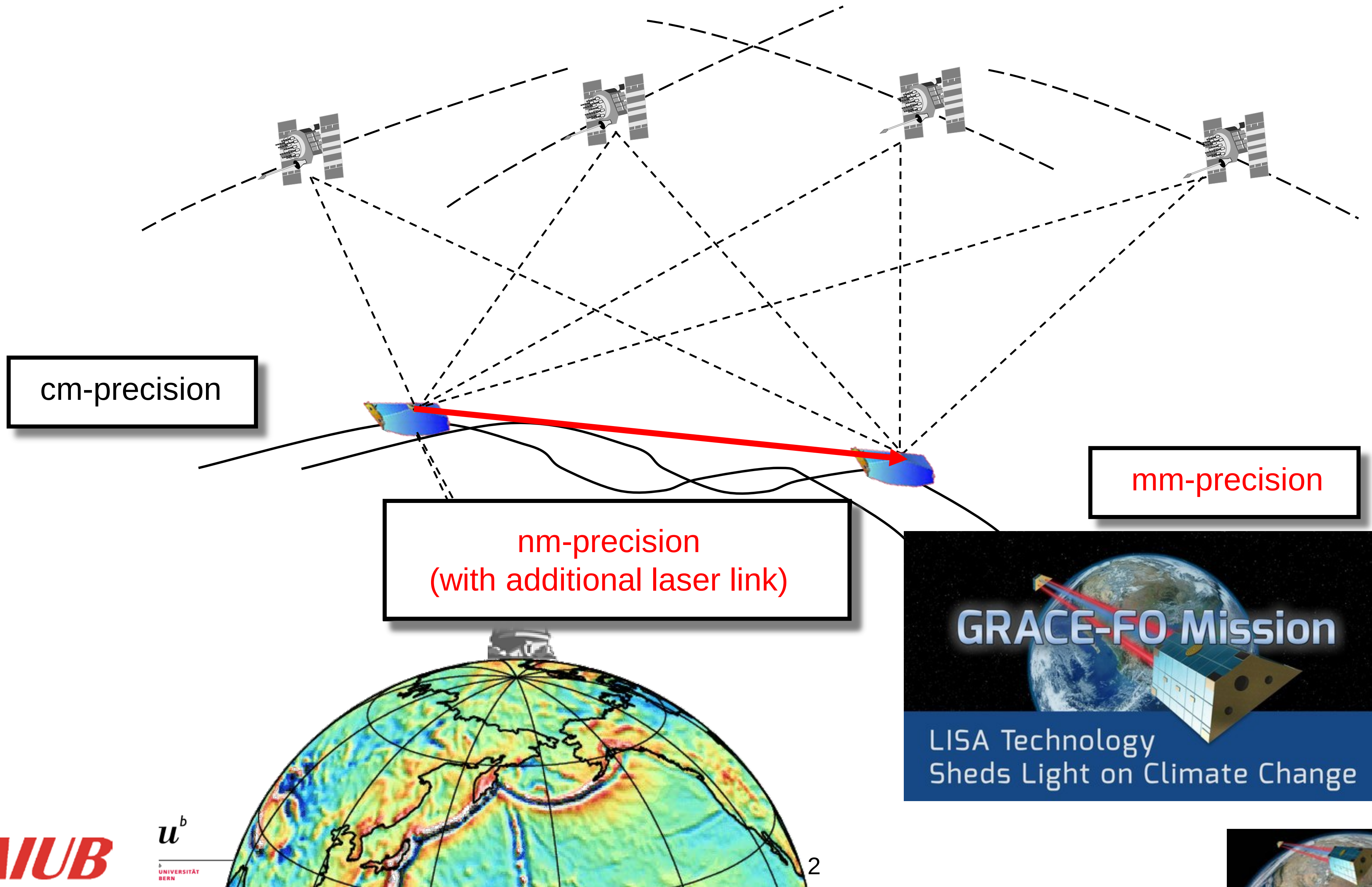
Lecture Contents

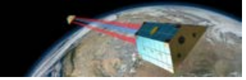
A multitude of Earth Observation Satellites





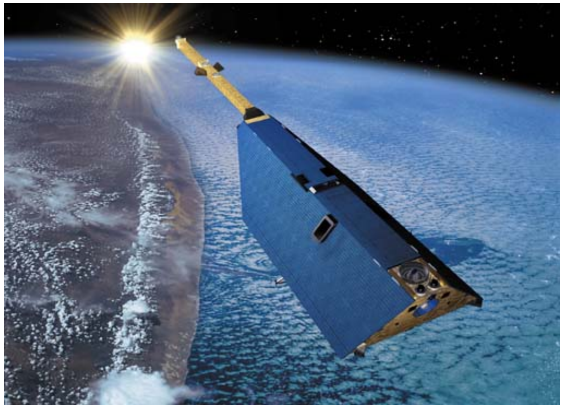
Precise Tracking Data in Near Earth Space





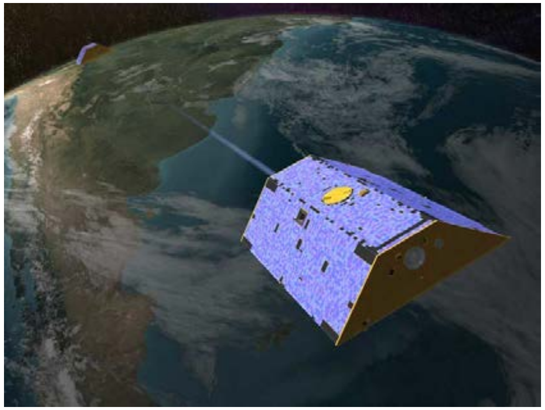
Low Earth Orbiters (LEOs)

CHAMP



CHAllenging **M**inisatellite **P**ayload

GRACE



Gravity **R**ecovery **A**nd **C**limate **E**xperiment

GOCE

Gravity and steady-state **O**cean **C**irculation **E**xplorer

Of course, there are many more missions equipped with GPS receivers

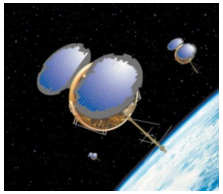
Jason

Jason-2

MetOp-A

Icesat

COSMIC



LEO Constellations

TanDEM-X

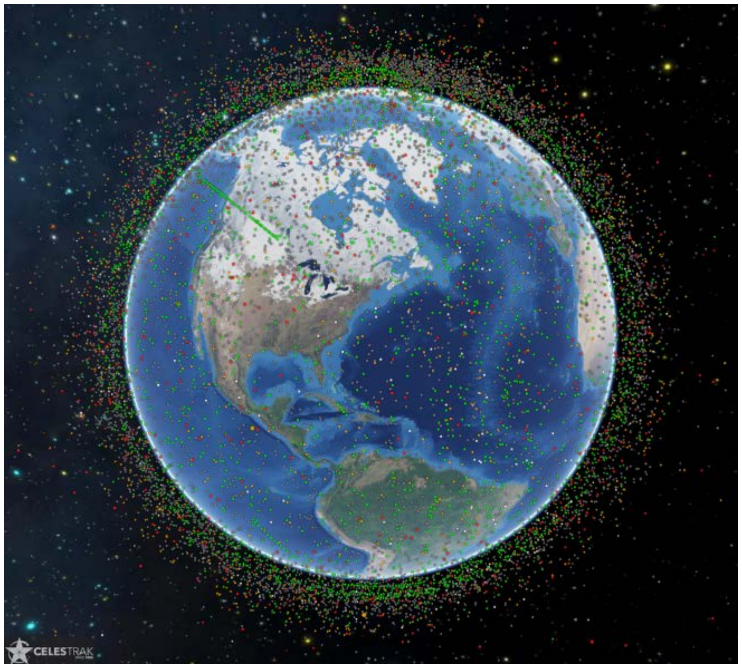
Swarm

Sentinel

GRACE-FO

COSMIC-2

LEO CubeSats



A multitude of **CubeSats** already exist or are planned for the future

e.g.

Spire

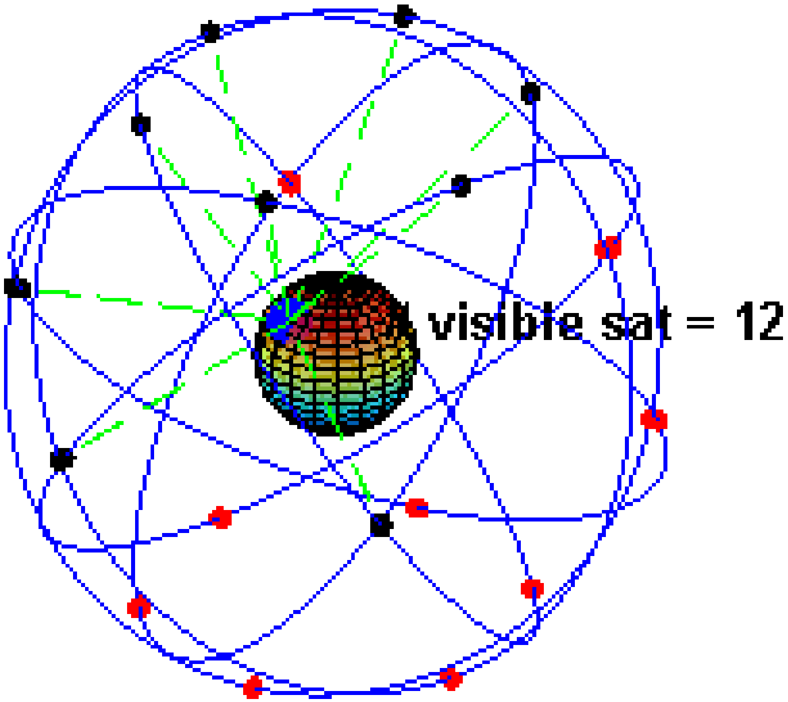
~ 100 Spire satellites are already in different orbits, offering dual-frequency high-quality GPS data

Many more will follow …

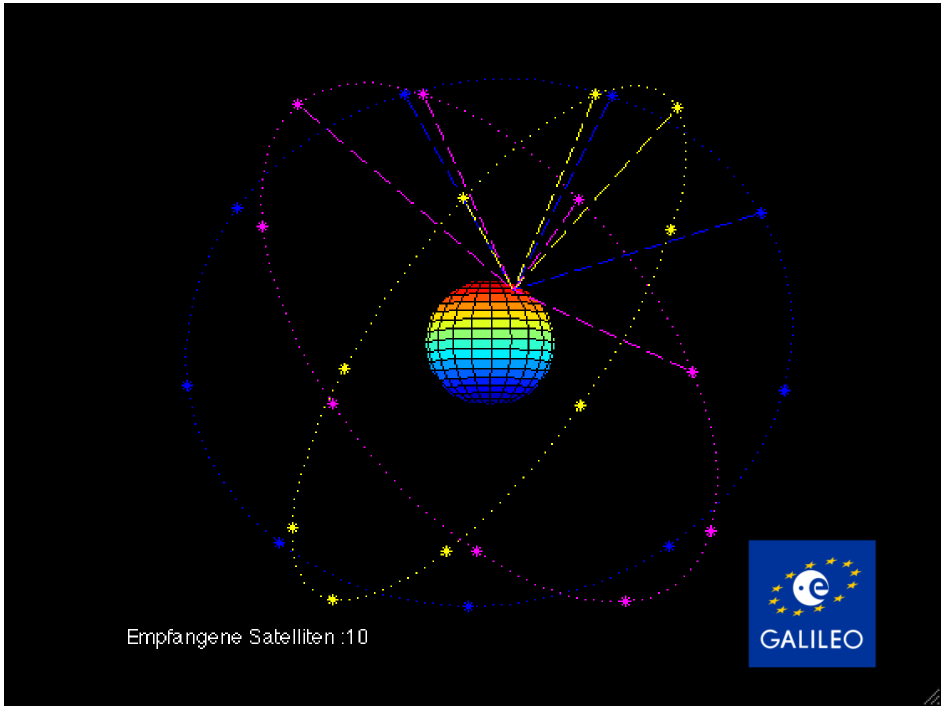
Global Positioning System

Introduction to GPS

GPS



Galileo



Other **Global Navigation Satellite Systems** (GNSS) are also available (GLONASS, Galileo, Beidou), but for a long time no multi-GNSS spaceborne receivers were in orbit. This changed with the launches of Fengyun-3, COSMIC-2, Sentinel-6.

Introduction to GPS

GPS: **G**lobal **P**ositioning **S**ystem

Characteristics:

- Satellite system for (real-time) **Positioning** and **Navigation**
- **Global** (everywhere on Earth, up to altitudes of 5000km) and **at any time**
- **Unlimited** number of users
- **Weather-independent** (radio signals are passing through the atmosphere)
- **3-dimensional position**, **velocity** and **time** information

GPS Segments

The GPS consists of **3 main segments**:

- **Space Segment**: the satellites and the constellation of satellites

- **Control Segment**: the ground stations, infrastructure and software for operation and monitoring of the GPS

- **User Segment**: all GPS receivers worldwide and the corresponding processing software

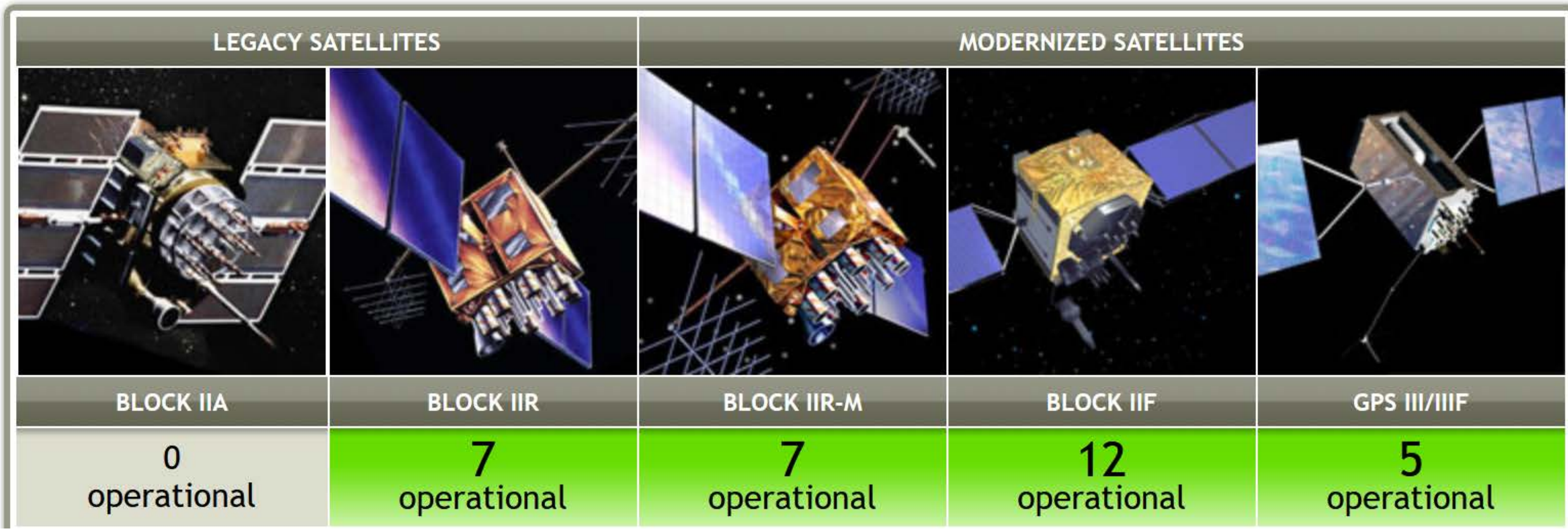
We should add an important **4th segment**:

- **Ground Segment**: all civilian permanent networks of reference sites and the international/regional/local services delivering products for the users

Space Segment

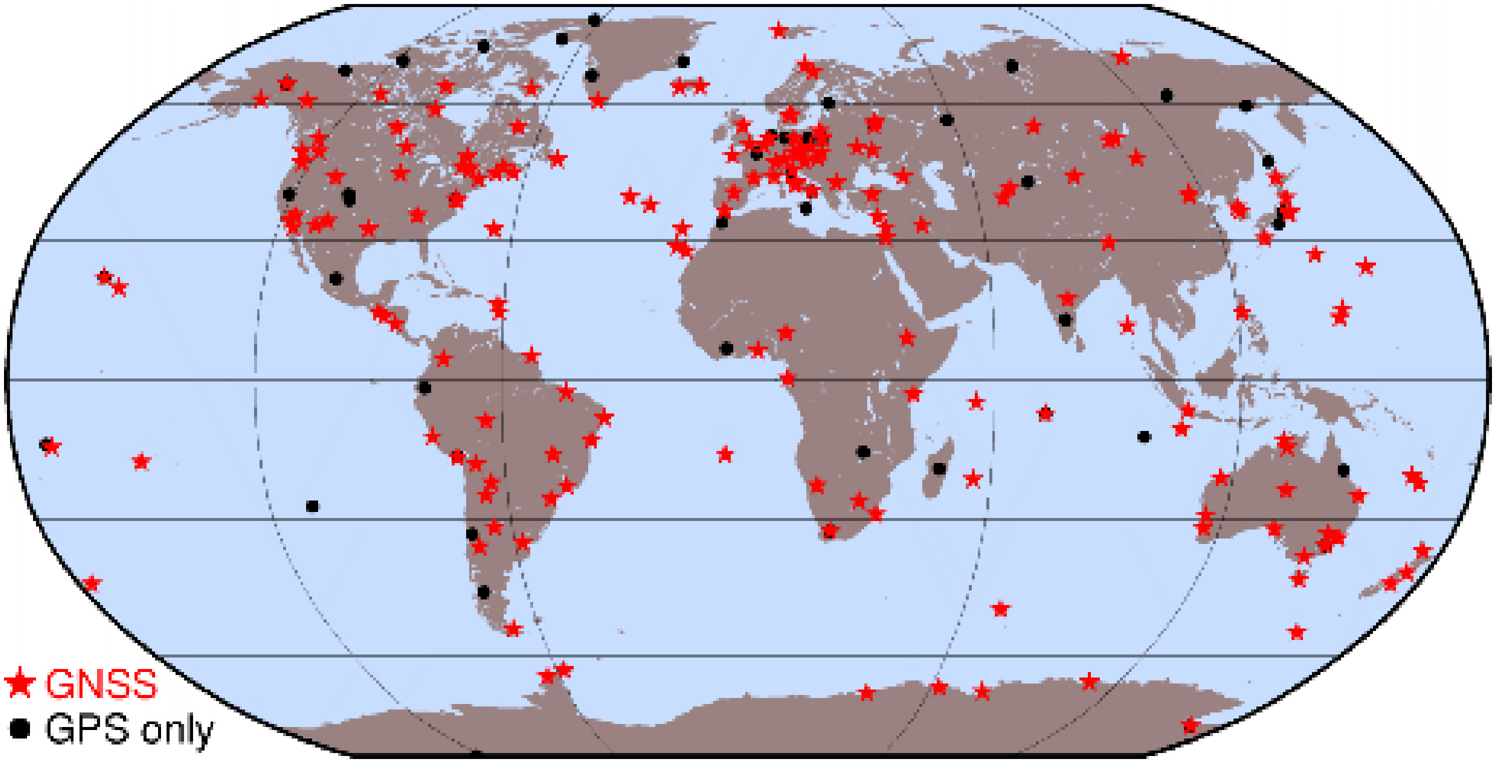
- The space segment nominally consists of **24 satellites**, presently: 31 active GPS satellites

- Constellation design: at least **4 satellites** in view from **any location** on the Earth at **any time**



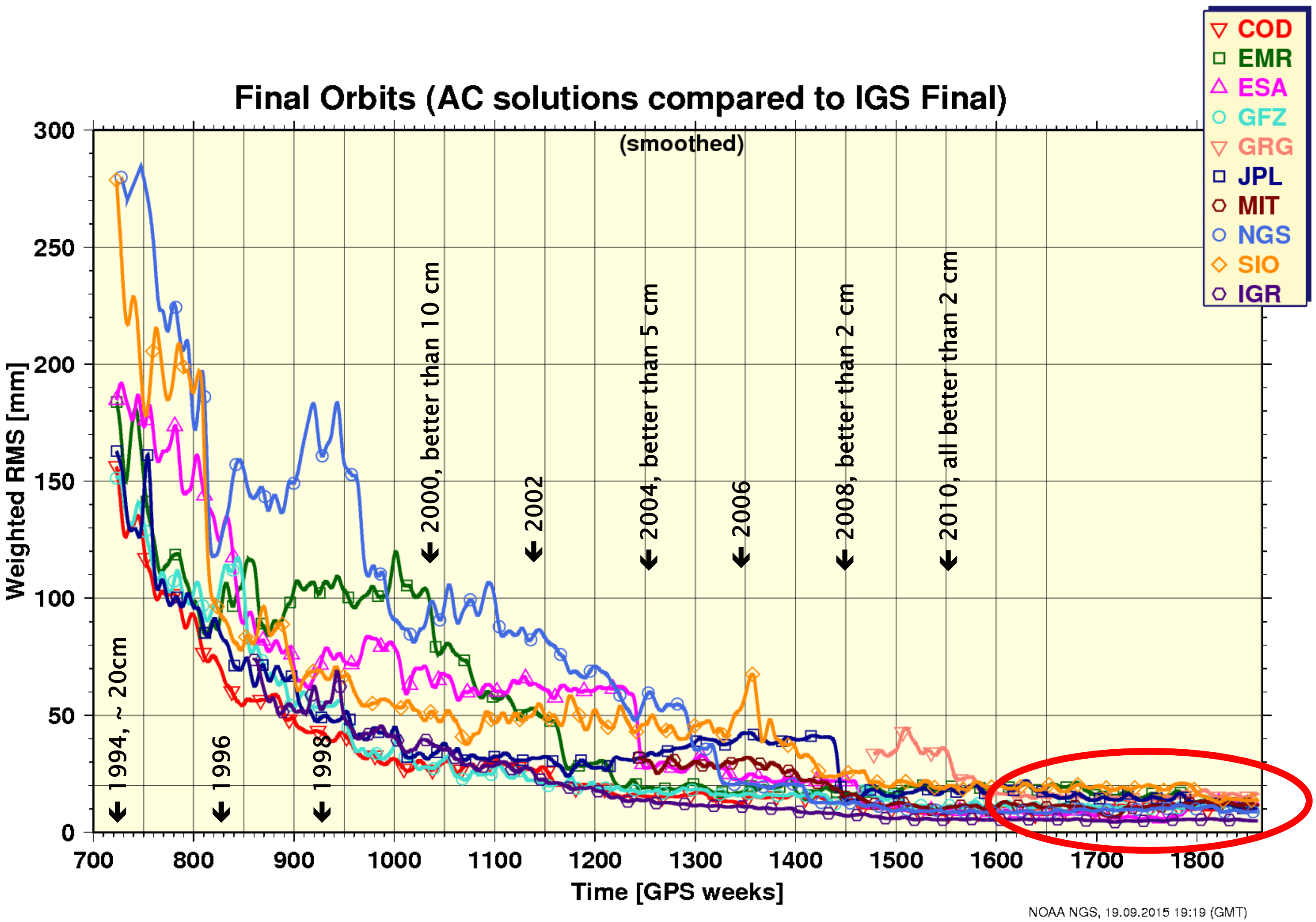
https://www.gps.gov/systems/gps/space/

Global Network of the IGS



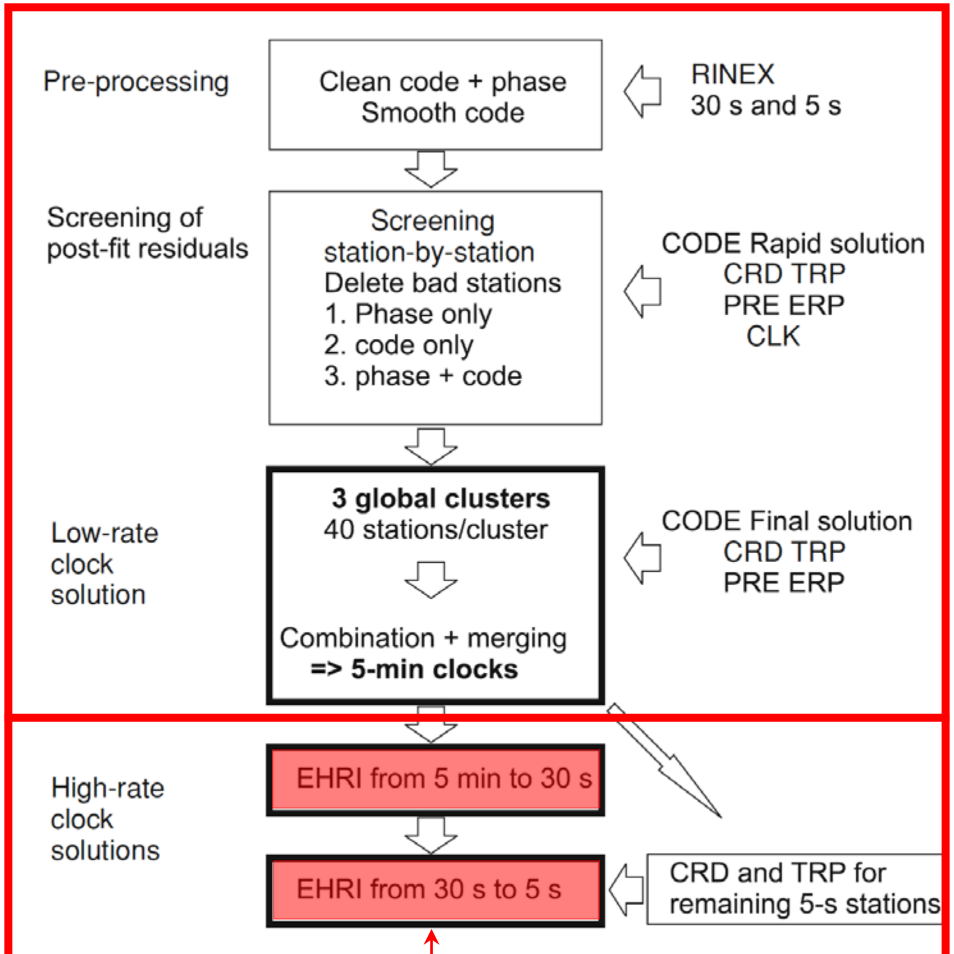
IGS stations used for computation of final orbits at CODE (Dach et al., 2009)

Performance of IGS Final Orbits



http://acc.igs.org/

Computation of High-Rate Clocks



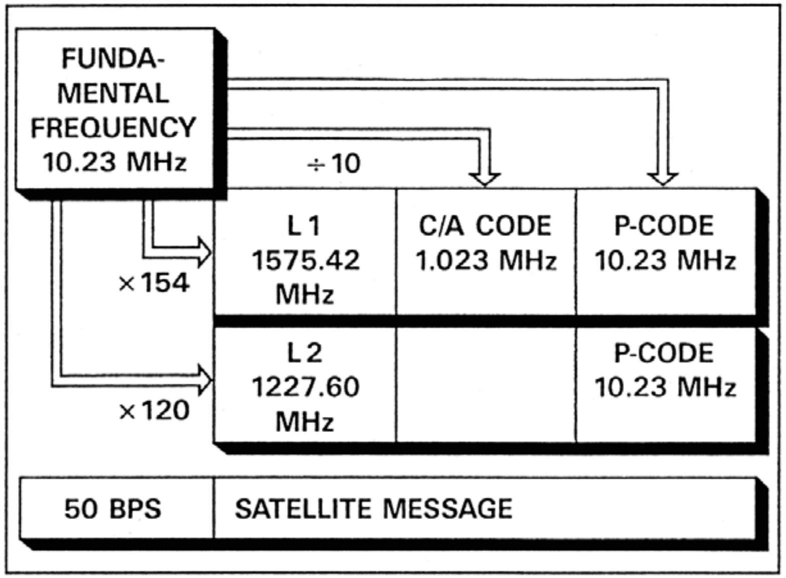
(Bock et al., 2009)

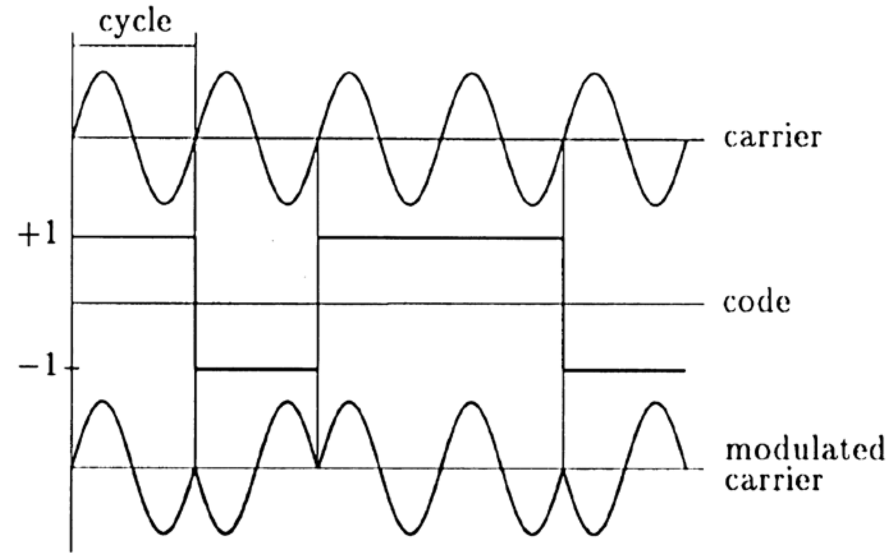
The final clock product with 5 min sampling is based on undifferenced GPS data of typically 120 stations of the IGS network

The IGS 1 Hz network is finally used for clock densification to 5 sec

The 5 sec clocks are interpolated to 1 sec as needed for 1 Hz kinematic LEO POD

GPS Signals





Signals driven by an **atomic clock**

Two **carrier signals** (sine waves):

- **L1**: f = 1575.43 MHz, λ = 19 cm
- **L2**: f = 1227.60 MHz, λ = 24 cm

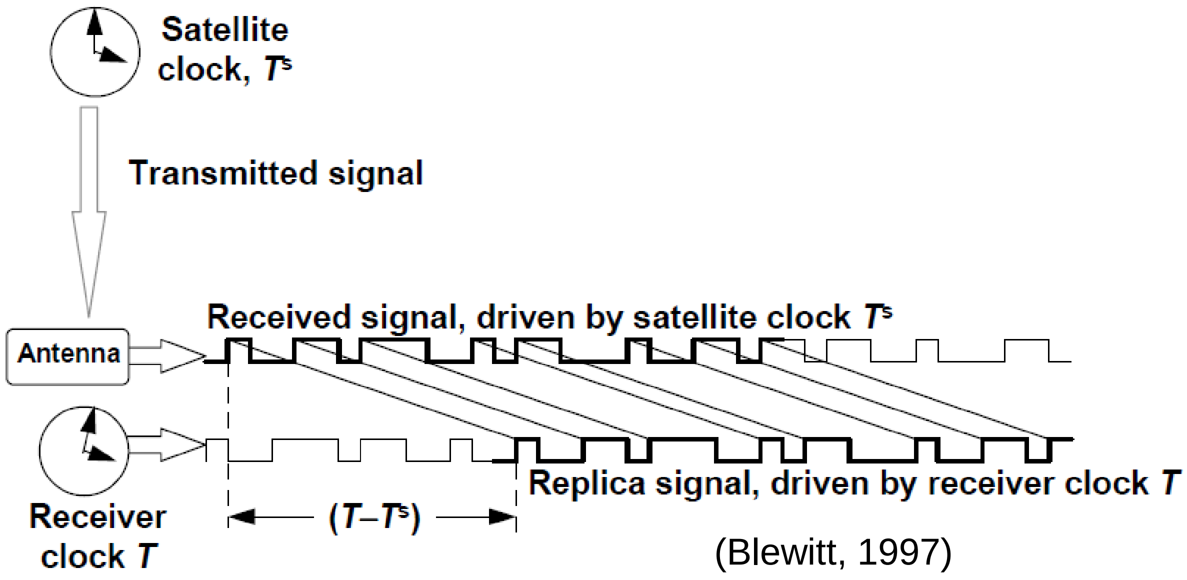
Bits encoded on carrier by phase modulation:

- **C/A-code** (Clear Access / Coarse Acquisition)
- **P-code** (Protected / Precise)
- **Broadcast/Navigation Message**

Pseudorange / Code Measurements

Code Observations P_i^k are defined as:

$$P_i^k \doteq c\,(T_i - T^k)$$



(Blewitt, 1997)

(Teunissen & Montenbruck, 2017)

Code Observation Equation

$$P_i^k = \rho_i^k - c \cdot \Delta t^k + c \cdot \Delta t_i$$

t_i, t^k GPS time of reception and emission

Δt^k Satellite clock offset $T^k - t^k$

Δt_i Receiver clock offset $T_i - t_i$

ρ_i^k Distance between receiver and satellite $c\,(t_i - t^k)$

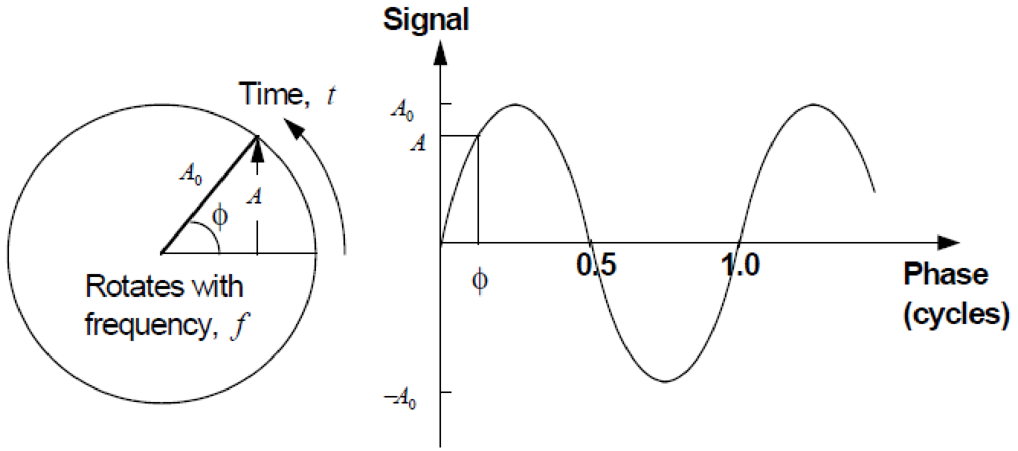
Known from ACs or IGS:

- satellite positions $(x^{k_j}, y^{k_j}, z^{k_j})$
- satellite clock offsets Δt^{k_j}

4 unknown parameters:

- receiver position (x_i, y_i, z_i)
- receiver clock offset Δt_i

Carrier Phase Measurements (1)



Phase ϕ (in cycles) increases linearly with time t:

$$\phi = f \cdot t$$

where f is the frequency

The **satellite** generates with its clock the phase signal ϕ^k. At emission time T^k (in satellite clock time) we have

$$\phi^k = f \cdot T^k$$

The same phase signal, e.g., a wave crest, propagates from the satellite to the receiver, but the receiver measures only the fractional part of the phase and does not know the **integer number of cycles** N_i^k (phase ambiguity):

$$\phi_i^k = \phi^k - N_i^k = f \cdot T^k - N_i^k$$

Carrier Phase Measurements (2)

The **receiver** generates with its clock a **reference phase.** At time of reception T_i of the satellite phase ϕ_i^k (in receiver clock time) we have:

$$\phi_i = f \cdot T_i$$

The actual **phase measurement** is the difference between receiver reference phase ϕ_i and satellite phase ϕ_i^k:

$$\psi_i^k = \phi_i - \phi_i^k = f \cdot T_i - (f \cdot T^k - N_i^k) = f \cdot (T_i - T^k) + N_i^k$$

Multiplication with the wavelength $\lambda = c/f$ leads to the **phase observation equation** in meters:

$$\begin{aligned} L_i^k &= \lambda \cdot \psi_i^k = c \cdot (T_i - T^k) + \lambda \cdot N_i^k \\ &= \rho_i^k - c \cdot \Delta t^k + c \cdot \Delta t_i + \lambda \cdot N_i^k \end{aligned}$$

Difference to the pseudorange observation: **integer ambiguity term** N_i^k

Detailed Observation Equation

$$L_i^k = \rho_i^k - c \cdot \Delta t^k + c \cdot \Delta t_i + \cancel{T_i^k} + \cancel{I_i^k} + \lambda \cdot N_i^k$$

$$+ \Delta_{rel} - c \cdot b^k + c \cdot b_i + m_i^k + \epsilon_i^k$$

ρ_i^k	Distance between satellite and receiver
Δt^k	Satellite clock offset wrt GPS time
Δt_i	Receiver clock offset wrt GPS time
$\cancel{T_i^k}$	~~Tropospheric delay~~
$\cancel{I_i^k}$	~~Ionospheric delay~~
N_i^k	Phase ambiguity
Δ_{rel}	Relativistic corrections
b^k	Delays in satellite (cables, electronics)
b_i	Delays in receiver and antenna
m_i^k	Multipath, scattering, bending effects
ϵ_i^k	Measurement error

Satellite positions and clocks are known from the IGS

Not existent for LEOs

Cancels out (first order only) when forming the ionosphere-free linear combination:

$$L_c = \frac{f_1^2}{f_1^2 - f_2^2} L_1 - \frac{f_2^2}{f_1^2 - f_2^2} L_2$$

Geometric Distance

Geometric distance ρ_{leo}^{k} is given by:

$$\rho_{leo}^{k} = \left|\boldsymbol{r}_{leo}(t_{leo}) - \boldsymbol{r}^{k}(t_{leo} - \tau_{leo}^{k})\right|$$

$\boldsymbol{r}_{leo}$	Inertial position of LEO antenna phase center at reception time
$\boldsymbol{r}^{k}$	Inertial position of GPS antenna phase center of satellite k at emission time
τ_{leo}^{k}	Signal traveling time between the two phase center positions

Different ways to represent $\boldsymbol{r}_{leo}$:

- **Kinematic** orbit representation
- **Dynamic** or **reduced-dynamic** orbit representation

Different Orbit Representations

Kinematic Orbit Representation (1)

Satellite position $\boldsymbol{r}_{leo}(t_{leo})$ (in inertial frame) is given by:

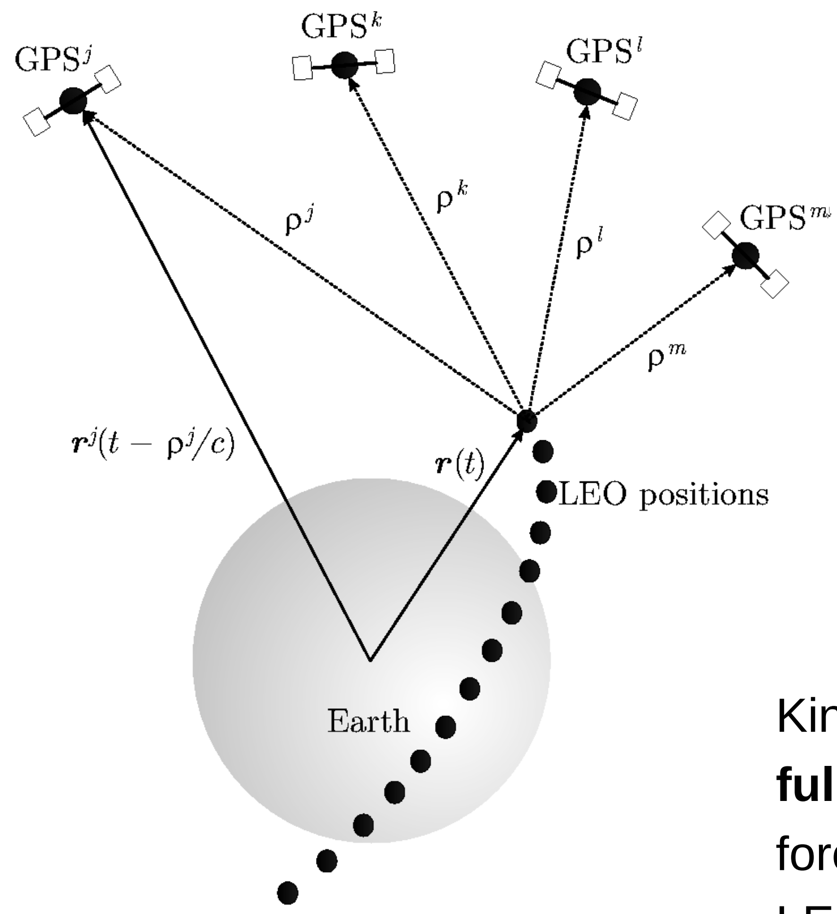
$$\boldsymbol{r}_{leo}(t_{leo}) = \boldsymbol{R}(t_{leo}) \cdot (\boldsymbol{r}_{leo,e,0}(t_{leo}) + \delta\boldsymbol{r}_{leo,e,ant}(t_{leo}))$$

$\boldsymbol{R}$	Transformation matrix from Earth-fixed to inertial frame
$\boldsymbol{r}_{leo,e,0}$	LEO center of mass position in Earth-fixed frame
$\delta\boldsymbol{r}_{leo,e,ant}$	LEO antenna phase center offset in Earth-fixed frame

Kinematic positions $\boldsymbol{r}_{leo,e,0}$ are estimated for each **measurement epoch**:

- Measurement epochs **need not** to be identical with nominal epochs
- Positions are **independent** of models describing the LEO dynamics. Velocities and accelerations cannot be provided in a "strict" sense.

Kinematic Orbit Representation (2)



A kinematic orbit is an ephemeris at **discrete** measurement epochs

Kinematic positions are **fully independent** on the force models used for LEO orbit determination

(Svehla and Rothacher, 2004)

Kinematic Orbit Representation (3)

Measurement epochs
(in GPS time)

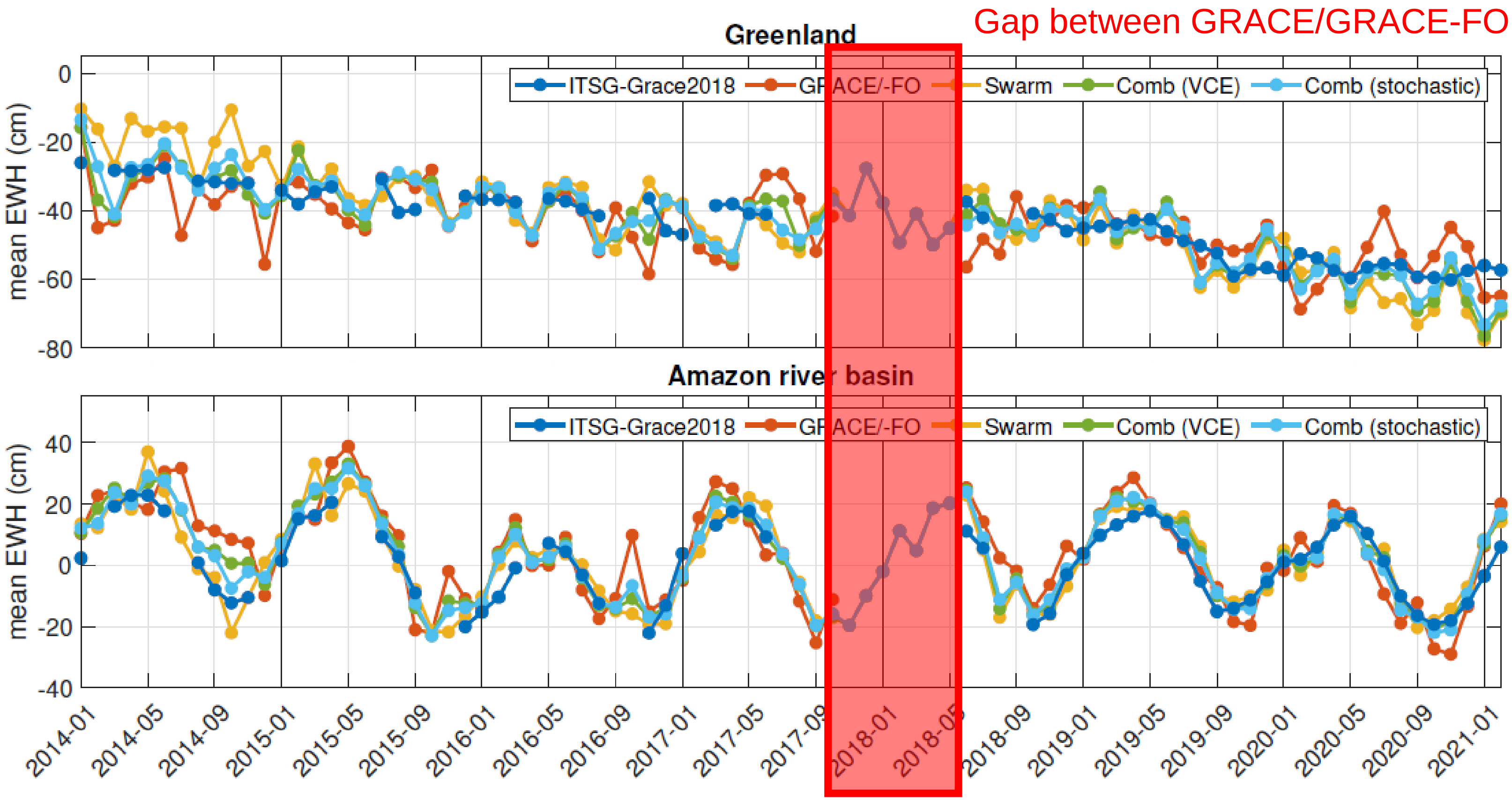
Positions (km)
(Earth-fixed)

```
*  2009 11  2  0  0  0.80678020
PL15  -390.612059   6623.987679     73.104149  193219.797196
*  2009 11  2  0  0  1.80678020
PL15  -389.240315   6624.166512     65.402457  193219.799413
*  2009 11  2  0  0  2.80678020
PL15  -387.868014   6624.336133     57.700679  193219.801634
*  2009 11  2  0  0  3.80678020
PL15  -386.495163   6624.496541     49.998817  193219.803855
*  2009 11  2  0  0  4.80678019
PL15  -385.121760   6624.647724     42.296889  193219.806059
*  2009 11  2  0  0  5.80678019
PL15  -383.747819   6624.789703     34.594896  193219.808280
*  2009 11  2  0  0  6.80678019
PL15  -382.373332   6624.922464     26.892861  193219.810495
*  2009 11  2  0  0  7.80678019
PL15  -380.998306   6625.046003     19.190792  193219.812692
*  2009 11  2  0  0  8.80678019
PL15  -379.622745   6625.160329     11.488692  193219.814899
*  2009 11  2  0  0  9.80678018
PL15  -378.246651   6625.265448      3.786580  193219.817123
```

Clock correction to nominal epoch (μs), e.g., to epoch 00:00:03

Excerpt of kinematic GOCE positions at begin of 2 Nov, 2009

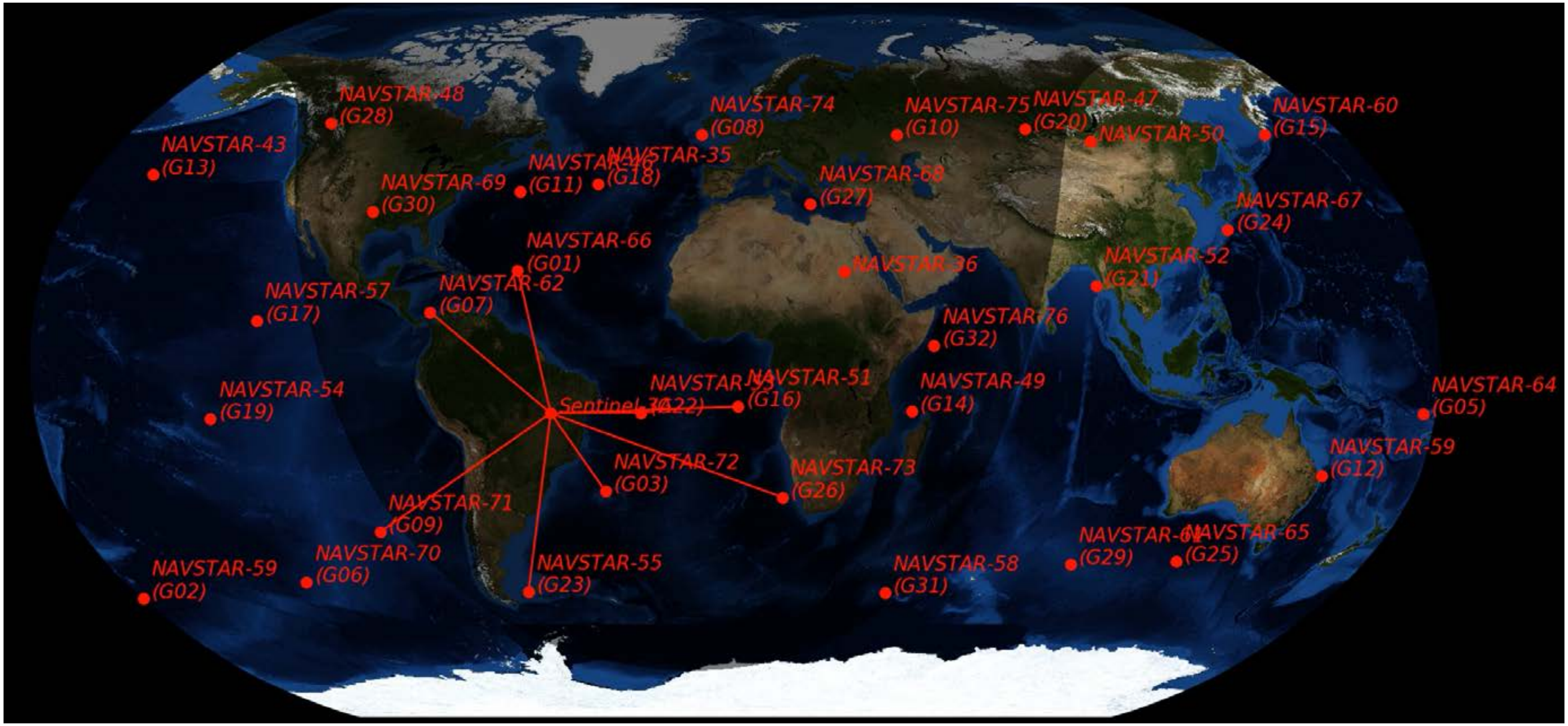
Recovery of Large-Scale Time-Variable Gravity Signals



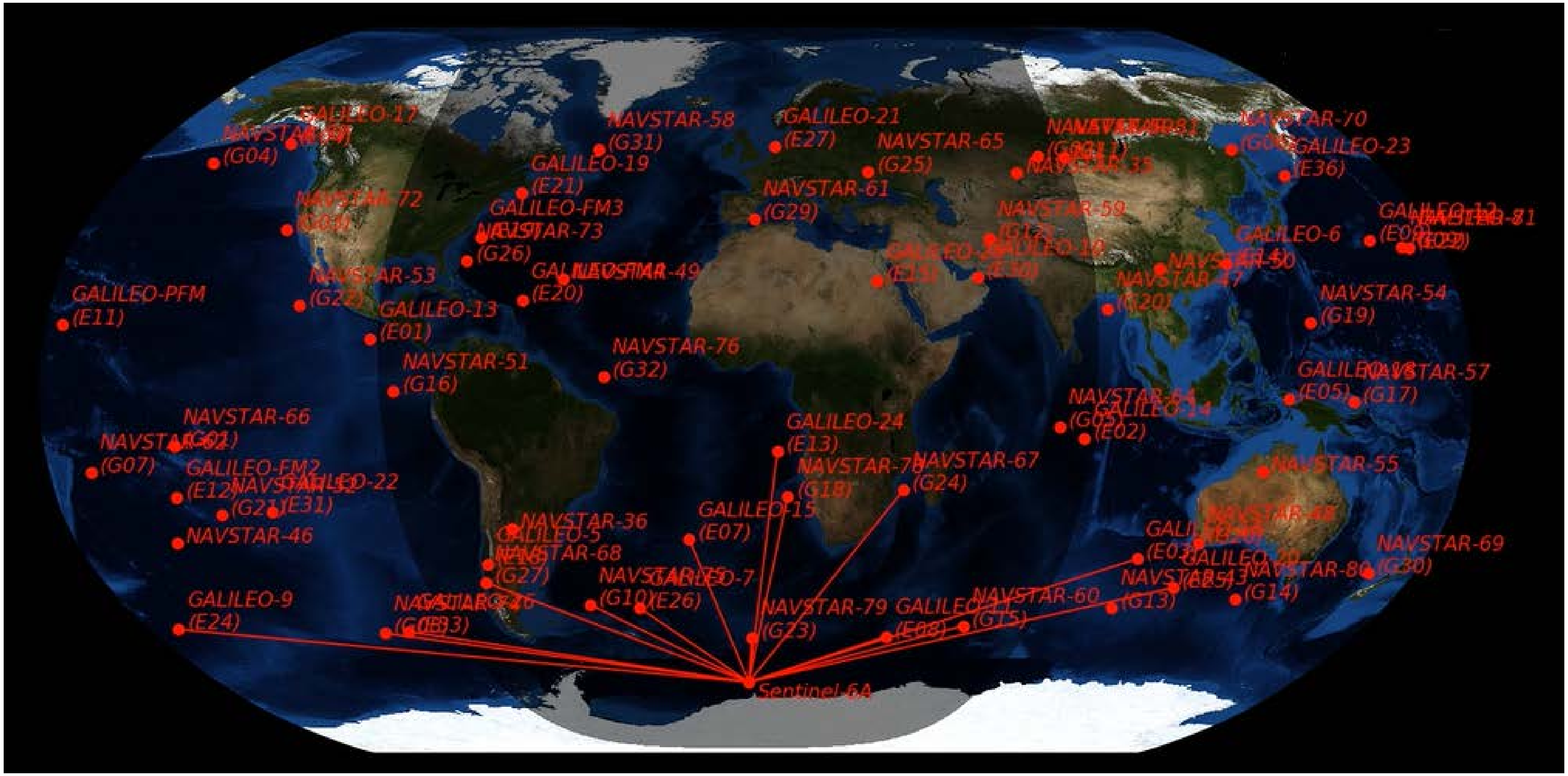
Kinematic positions allow it to recover the **long wavelength** part of the Earth's time-variable gravity field. The scatter of monthly gravity field solutions is larger than from dedicated GRACE/GRACE-FO data, but trends and annual signals may be derived remarkably well.

(Grombein et al., 2022)

Example: Sentinel-3A GPS Tracking



Example: Sentinel-6A multi-GNSS Tracking



Simulated Data Set (1)

The simplified **Code Observation Equation** of the simulation reads as

$$\bar{\rho}_{k,i} = \sqrt{(x_{k,i} - x_{leo,i})^2 + (y_{k,i} - y_{leo,i})^2 + (z_{k,i} - z_{leo,i})^2} + c_{leo,i} \quad , \quad k = 1, \ldots, n_{sat}$$

with

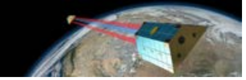
$(x_{k,i}, y_{k,i}, z_{k,i})$	Known inertial position of GPS satellite *k* at epoch *i*
$(x_{leo,i}, y_{leo,i}, z_{leo,i})$	Unknown inertial LEO position at epoch *i*
$c_{leo,i}$	Unknown clock correction of LEO receiver at epoch *i*

The simplified **Phase Observation Equation** of the simulation reads as

$$\bar{\lambda}_{k,i} = \sqrt{(x_{k,i} - x_{leo,i})^2 + (y_{k,i} - y_{leo,i})^2 + (z_{k,i} - z_{leo,i})^2} + c_{leo,i} + b_k \quad , \quad k = 1, \ldots, n_{sat}$$

with the additional parameter

b_k	Unknown constant phase bias to the satellite *k*



Simulated Data Set (2)

Tabelle 1: Code observations $\rho'_{k,i}$ contained in the file `OBS_CODE.txt`. They are stored in `template.m` in the array `obs_code(kepo,isat)`. In analogy the phase observations are contained in the file `OBS_PHASE.txt` and are stored in the array `obs_phase(kepo,isat)`. The observation times are also computed in the source code by `t = 10*(kepo-1)`, kepo $= 1, \ldots,$ nepo.

Time t_i (sec)	Sat. 1 (m)	Sat. 2 (m)	Sat. 3 (m)	Sat. 4 (m)	Sat. 5 (m)	etc.
0.0000	19447557.3266	0.0000	0.0000	0.0000	21654965.4010	...
10.0000	19446601.2222	0.0000	0.0000	0.0000	21678374.5718	...
20.0000	19446475.1291	0.0000	0.0000	0.0000	21702249.7103	...
...	...	...	...	...	....	...

Tabelle 2: Positions $(x_{k,i}, y_{k,i}, z_{k,i})$ of the GPS satellites contained in the file `GPS.txt`. They are stored in the array `r_gps(coord,iepo,ksat)`.

x GPS 1 (m)	y GPS 1 (m)	z GPS 1 (m)	x GPS 2	y GPS 2	z GPS 2	etc.
26235000.0000	0.0000	0.0000	-789889.9012	-15196033.7951	-21702185.3748	...
26234971.0435	22468.8157	32088.7944	-751120.9506	-15196458.0361	-21702791.2539	...
26234884.1742	44937.5818	64177.5179	-712350.3922	-15196849.7458	-21703350.6732	...
...	...	...	...	...	....	...

Simulated Data Set (3)

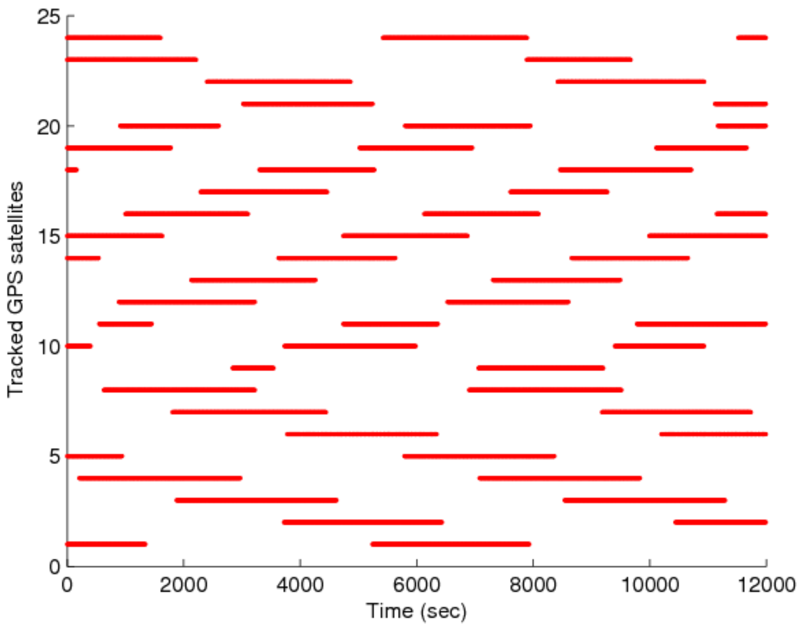
Tabelle 3: Positions $(x_{leo,i}, y_{leo,i}, z_{leo,i})$ and velocities of the LEO satellite contained in the file `LEO.txt`. The positions are stored in the array `r_leo(coord,iepo)`. The velocities are not needed for this project unless the orbit differences shall be plotted in a radial, along-track, cross-track frame instead of the inertial x,y,z frame.

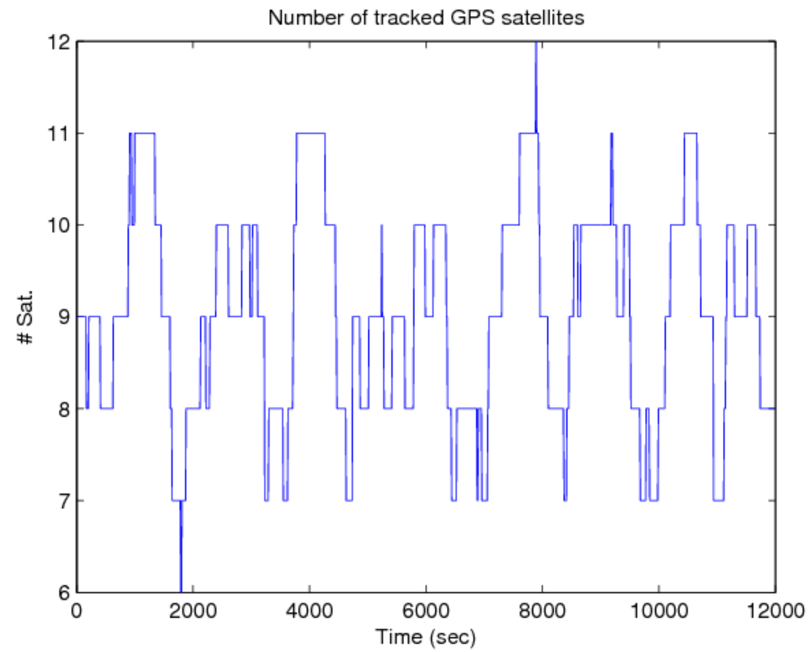
x LEO (m)	y LEO (m)	z LEO (m)	v_x LEO (m/s)	v_y LEO (m/s)	v_z LEO (m/s)
6824717.7284	1203381.8712	0.0000	-0.0000	0.0000	7621.8949
6824309.0437	1203309.8091	76217.4278	-81.7361	-14.4123	7621.4385
6823083.0403	1203093.6316	152425.7274	-163.4621	-28.8228	7620.0693
...	...	...	...	...	...

Tabelle 4: Biases b_k of the phase observations to satellite k contained in the file `BIASES.txt`. They are stored in the array `true_bias(ksat)`.

b_1 (m)	b_2 (m)	b_3 (m)	b_4 (m)	etc.
2319	-1149	2918	-10144	...

Simulated Data Set (4)





Tracking scenario of the simulated data set (left). Up to 12 GPS satellites are at maximum simultaneously visible from the LEO satellite (right). The viewing geometry is continuously changing due to the orbital motion of all satellites.

Dynamic Orbit Representation (1)

Satellite position $\boldsymbol{r}_{leo}(t_{leo})$ (in inertial frame) is given by:

$$\boldsymbol{r}_{leo}(t_{leo}) = \boldsymbol{r}_{leo,0}(t_{leo}; a, e, i, \Omega, \omega, u_0; Q_1, \ldots, Q_d) + \delta\boldsymbol{r}_{leo,ant}(t_{leo})$$

$\boldsymbol{r}_{leo,0}$	LEO center of mass position
$\delta\boldsymbol{r}_{leo,ant}$	LEO antenna phase center offset
$a, e, i, \Omega, \omega, u_0$	LEO initial osculating orbital elements
$Q_1, \ldots, Q_d$	LEO dynamical parameters

Satellite trajectory $\boldsymbol{r}_{leo,0}$ is a particular solution of an **equation of motion**

- One set of **initial conditions** (orbital elements) is estimated per arc. Dynamical parameters of the force model may be estimated on request.

Dynamic Orbit Representation (2)

Equation of motion (in inertial frame) is given by:

$$\ddot{\boldsymbol{r}} = -GM\frac{\boldsymbol{r}}{r^3} + \boldsymbol{f}_1(t, \boldsymbol{r}, \dot{\boldsymbol{r}}, Q_1, ..., Q_d)$$

with initial conditions

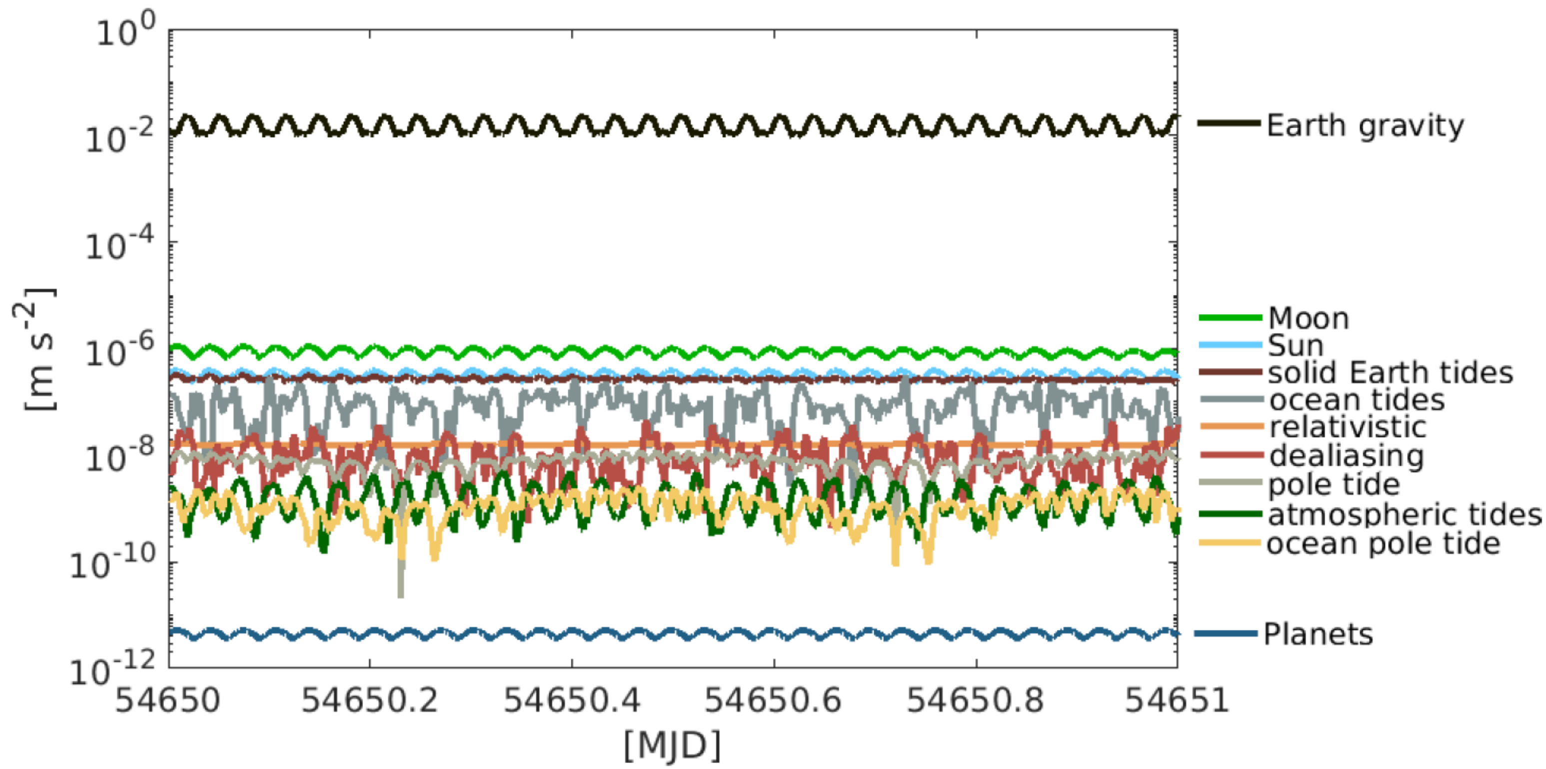
$$\boldsymbol{r}(t_0) = \boldsymbol{r}(a, e, i, \Omega, \omega, u_0; t_0)$$
$$\dot{\boldsymbol{r}}(t_0) = \dot{\boldsymbol{r}}(a, e, i, \Omega, \omega, u_0; t_0)$$

The **acceleration** $\boldsymbol{f}_1$ consists of **gravitational** and **non-gravitational** perturbations taken into account to model the satellite trajectory. Unknown parameters $Q_1, ..., Q_d$ of force models may appear in the equation of motion together with deterministic (known) accelerations given by analytical models.

Perturbing Accelerations of a LEO Satellite

Force	**Acceleration** (m/s²)
Central term of Earth's gravity field	8.42
Oblateness of Earth's gravity field	0.015
Atmospheric drag	0.00000079
Higher order terms of Earth's gravity field	0.00025
Attraction from the Moon	0.0000054
Attraction from the Sun	0.0000005
Direct solar radiation pressure	0.000000097
...	...

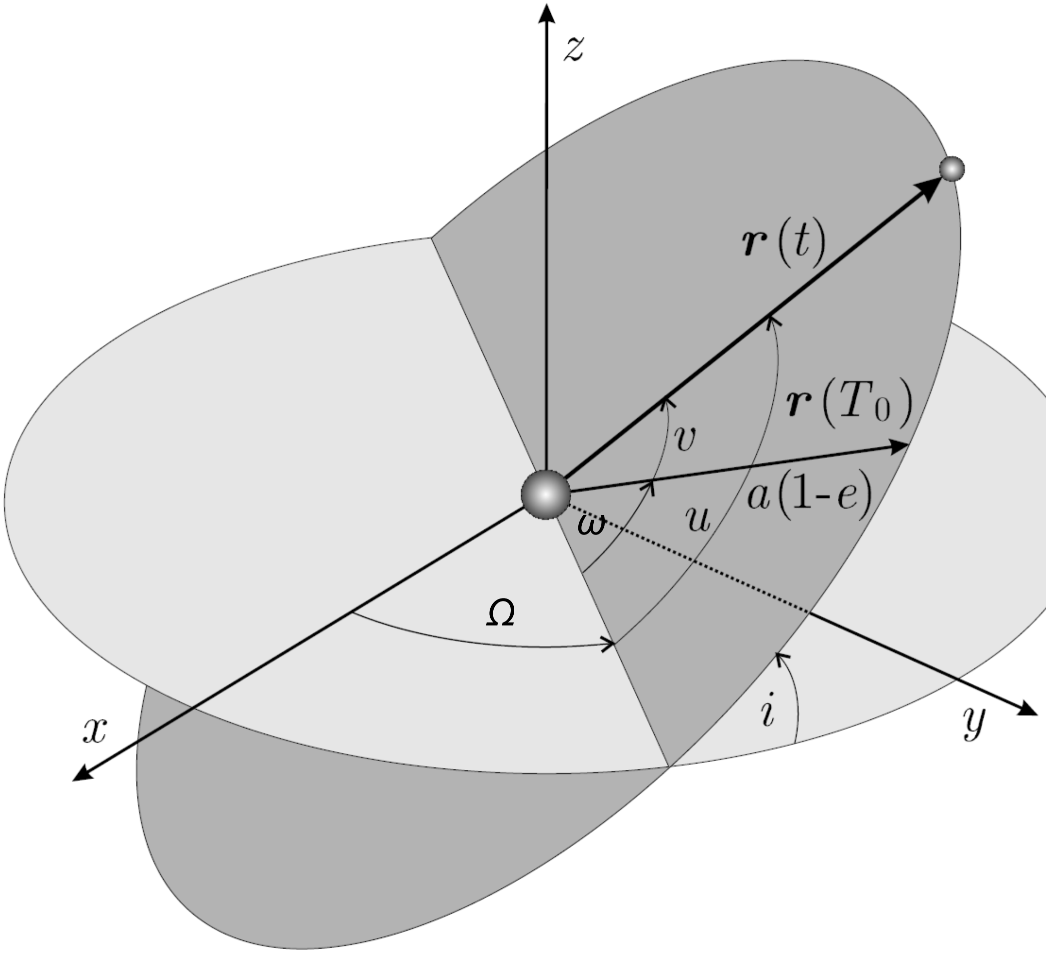
Perturbing Accelerations of a LEO Satellite



Norm of the **COST-G benchmark accelerations** along a GRACE satellite orbit. The benchmark data set may be used as a reference data set and provides the opportunity to test the implementation of corresponding background models.

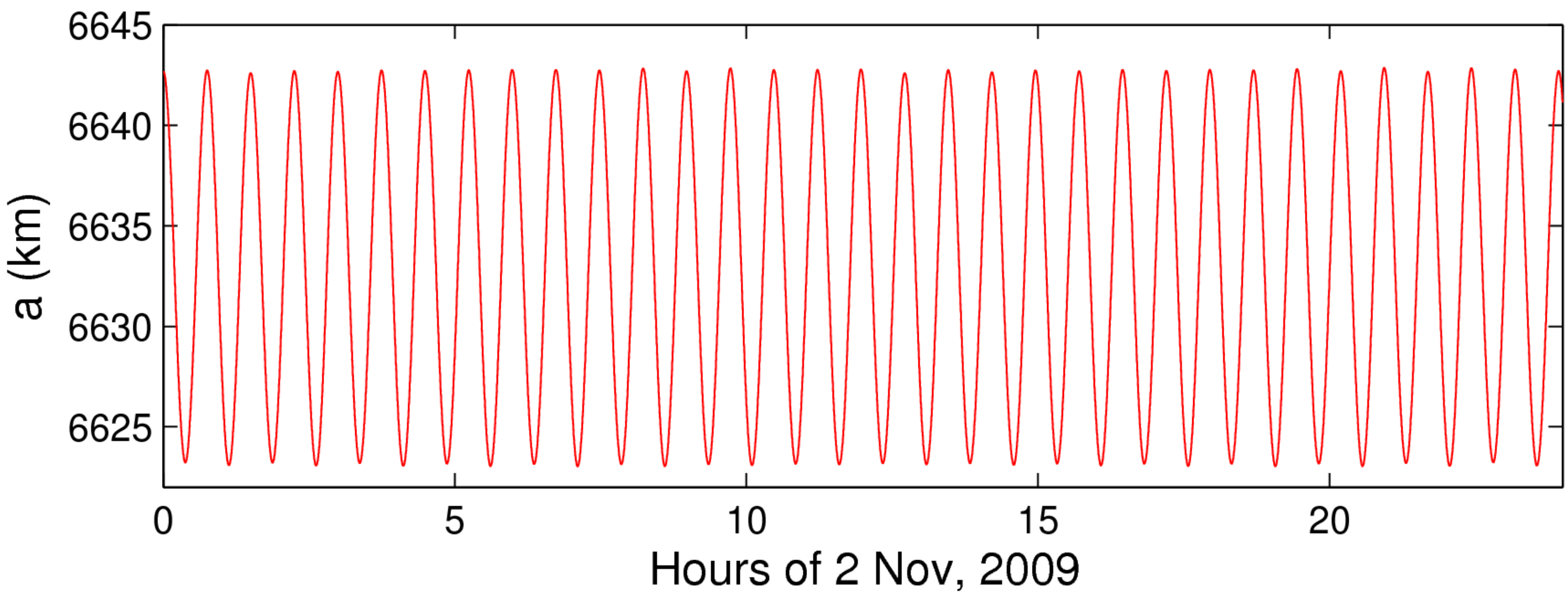
(Mayer-Gürr and Kvas, 2019; Lasser et al., 2020)

Osculating Orbital Elements



(Beutler, 2005)

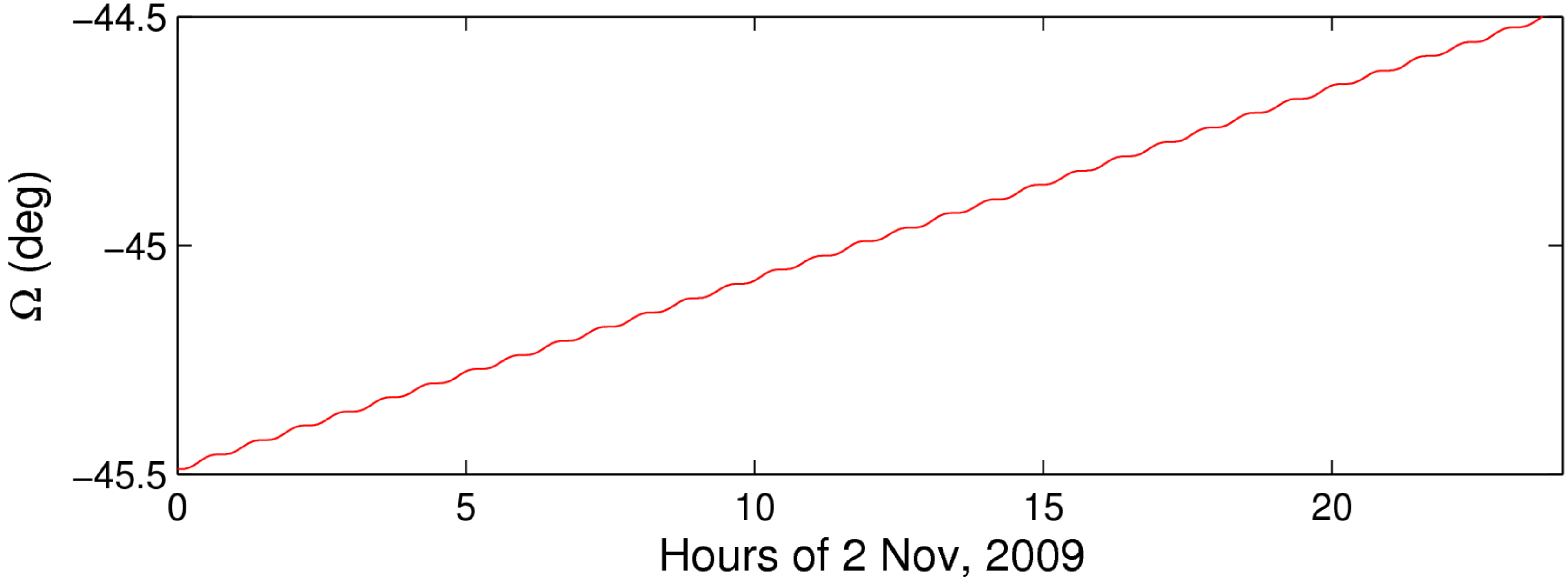
Osculating Orbital Elements of GOCE (1)



Semi-major axis:

Twice-per-revolution variations of about ±10 km around the mean semi-major axis of 6632.9km, which corresponds to a mean altitude of 254.9 km

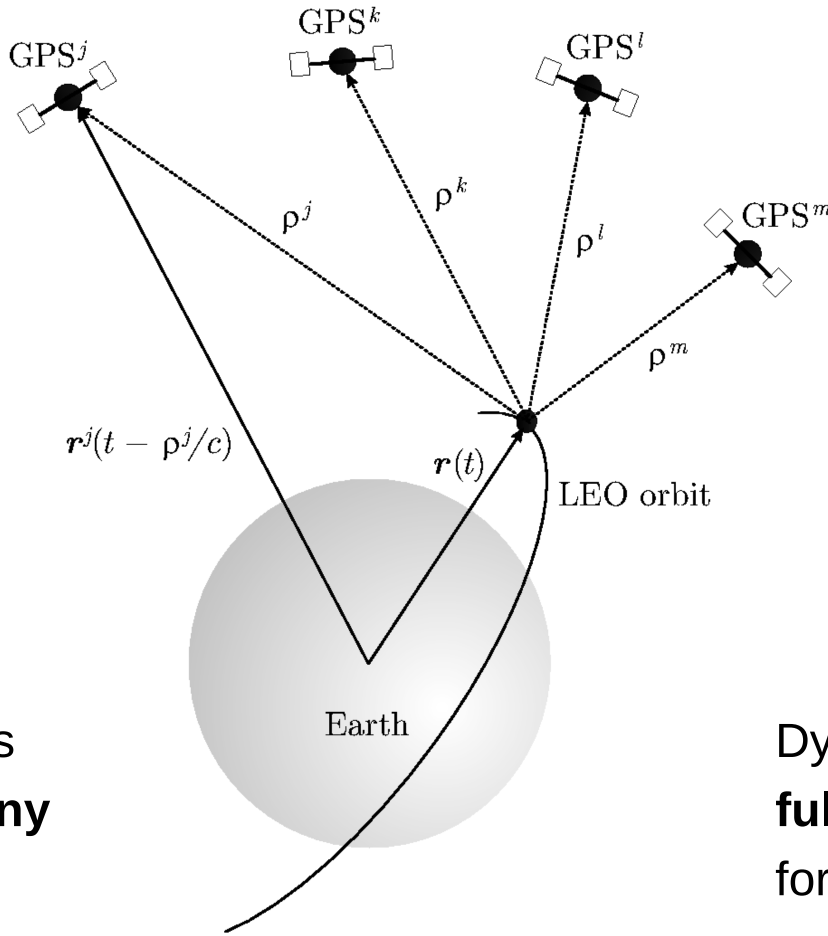
Osculating Orbital Elements of GOCE (2)



Right ascension of ascending node:

Twice-per-revolution variations and linear drift of about +1°/day (360°/365days) due to the sun-synchronous orbit

Dynamic Orbit Representation (3)



Dynamic orbit positions may be computed at **any epoch** within the arc

Dynamic positions are **fully dependent** on the force models used, e.g., on the gravity field model

Reduced-Dynamic Orbit Representation (1)

Equation of motion (in inertial frame) is given by:

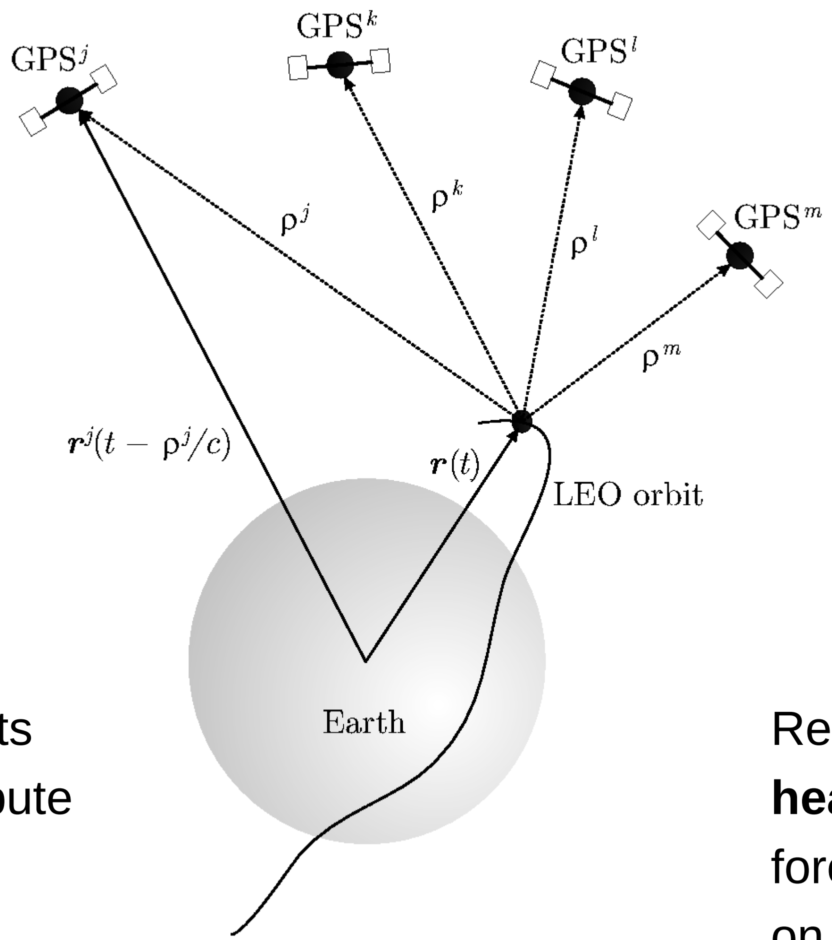
$$\ddot{\boldsymbol{r}} = -GM\frac{\boldsymbol{r}}{r^3} + \boldsymbol{f}_1(t, \boldsymbol{r}, \dot{\boldsymbol{r}}, Q_1, ..., Q_d, P_1, ..., P_s)$$

$P_1, ..., P_s$ Pseudo-stochastic parameters

Pseudo-stochastic parameters are:

- additional empirical parameters characterized by a priori known **statistical properties**, e.g., by expectation values and a priori variances
- useful to **compensate** for deficiencies in dynamic models, e.g., deficiencies in models describing non-gravitational accelerations
- often set up as **piecewise constant accelerations** to ensure that satellite trajectories are continuous and differentiable at any epoch

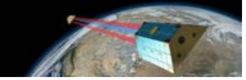
Reduced-Dynamic Orbit Representation (2)



Reduced-dynamic orbits are well suited to compute LEO orbits of **highest quality**

(Jäggi et al., 2006; Jäggi, 2007)

Reduced-dynamic orbits **heavily depend** on the force models used, e.g., on the gravity field model



Reduced-dynamic Orbit Representation (3)

Position epochs
(in GPS time)

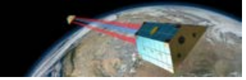
Positions (km) &
Velocities (dm/s)
(Earth-fixed)

```
*  2009 11  2  0  0  0.00000000
PL15   -391.718353   6623.836682     79.317661 999999.999999
VL15  13710.157683   1908.731015 -77015.601314 999999.999999
*  2009 11  2  0  0 10.00000000
PL15   -377.980705   6625.284690      2.298385 999999.999999
VL15  13764.602016    987.250587 -77021.193676 999999.999999
*  2009 11  2  0  0 20.00000000
PL15   -364.190222   6625.811136    -74.721213 999999.999999
VL15  13815.825127     65.631014 -77016.232293 999999.999999
*  2009 11  2  0  0 30.00000000
PL15   -350.350131   6625.415949   -151.730567 999999.999999
VL15  13863.820409   -855.995477 -77000.719734 999999.999999
*  2009 11  2  0  0 40.00000000
PL15   -336.463660   6624.099187   -228.719134 999999.999999
VL15  13908.581905  -1777.497047 -76974.660058 999999.999999
*  2009 11  2  0  0 50.00000000
PL15   -322.534047   6621.861041   -305.676371 999999.999999
VL15  13950.104280  -2698.741871 -76938.058807 999999.999999
*  2009 11  2  0  1  0.00000000
PL15   -308.564533   6618.701833   -382.591743 999999.999999
VL15  13988.382807  -3619.598277 -76890.923043 999999.999999
```

Clock corrections
are not provided

Excerpt of reduced-dynamic GOCE positions at begin of 2 Nov, 2009

Principles of Orbit Determination



Principle of Orbit Determination

The **actual orbit** $\boldsymbol{r}(t)$ is expressed as a truncated Taylor series:

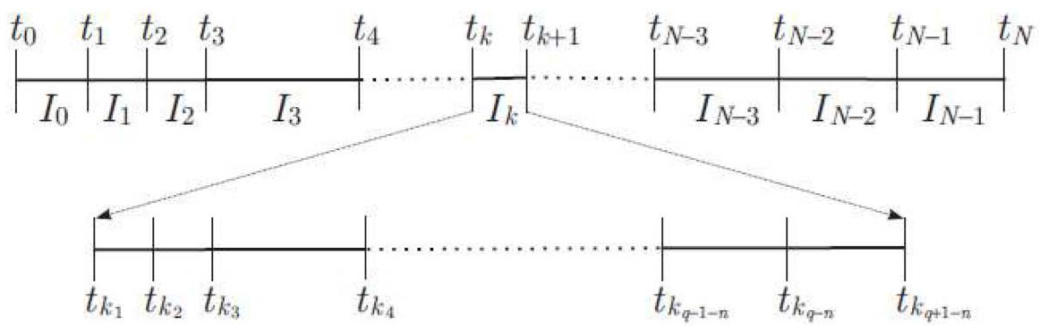
$$\boldsymbol{r}(t) = \boldsymbol{r}_0(t) + \sum_{i=1}^{n} \frac{\partial \boldsymbol{r}_0}{\partial P_i}(t) \cdot (P_i - P_{0,i})$$

$\boldsymbol{r}_0(t)$	A priori orbit
$\frac{\partial \boldsymbol{r}_0}{\partial P_i}(t)$	Partial derivative of the a priori orbit $\boldsymbol{r}_0(t)$ w.r.t. parameter P_i
$P_{0,i}$	A priori parameter values of the a priori orbit $\boldsymbol{r}_0(t)$
P_i	Parameter values of the improved orbit $\boldsymbol{r}(t)$

A **least-squares** adjustment of spacecraft tracking data yields **corrections** to the a priori parameter values $P_{0,i}$. Using the above equation, the improved (linearized) orbit $\boldsymbol{r}(t)$ may be eventually computed.

Numerical Integration (1)

Collocation algorithms (one particular class of numerical integration techniques) are subsequently used to briefly illustrate the principles of numerical integration:



The original intervall is divided into N integration intervals. For each interval I_k a further subdivision is performed according to the order q of the adopted method. At these points t_{k_j} the numerical solution is requested to solve the differential equation system of order n.

(Beutler, 2005)

Numerical Integration (2)

Initial value problem in the interval I_k is given by:

$$\ddot{\mathbf{r}}_k = \mathbf{f}(t, \mathbf{r}_k, \dot{\mathbf{r}}_k)$$

with initial conditions

$$\mathbf{r}_k(t_k) \doteq \mathbf{r}_{k0} \qquad \text{and} \qquad \dot{\mathbf{r}}_k(t_k) \doteq \dot{\mathbf{r}}_{k0}$$

where the initial values are defined as

$$\mathbf{r}_{k0}^{(i)} = \begin{cases} \mathbf{r}_0^{(i)} & ;\ k = 0 \\ \mathbf{r}_{k-1}^{(i)}(t_k) & ;\ k > 0 \end{cases}$$

Numerical Integration (3)

The **collocation method** approximates the solution in the interval I_k by:

$$\mathbf{r}_k(t) \doteq \sum_{l=0}^{q} \frac{1}{l!}(t - t_k)^l \, \mathbf{r}_{k0}^{(l)}$$

The coefficients $\mathbf{r}_{k0}^{(l)}, \; l = 0, \ldots, q$ are obtained by requesting that the numerical solution assumes the initial values and solves the differential equation system at $q-1$ different epochs $t_{k_j}, \; j = 1, \ldots, q-1$. This leads to the conditions

$$\sum_{l=2}^{q} \frac{(t_{k_j} - t_k)^{l-2}}{(l-2)!} \mathbf{r}_{k0}^{(l)} = \mathbf{f}(t_{k_j}, \mathbf{r}_k(t_{k_j}), \dot{\mathbf{r}}_k(t_{k_j})) \;, \quad j = 1, \ldots, q-1 \,.$$

They are non-linear but can be solved efficiently by an iterative procedure.

(Beutler, 2005)

Partial Derivatives

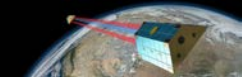
The partial of the r -th observation w.r.t. orbit parameter P_i may be expressed as

$$\frac{\partial F_r(\boldsymbol{X})}{\partial P_i} = \left(\boldsymbol{\nabla}\left(F_r(\boldsymbol{X})\right)\right)^T \cdot \ \frac{\partial \boldsymbol{r}_0}{\partial P_i}(t)$$

with the gradient given by

$$\left(\boldsymbol{\nabla}\left(F_r(\boldsymbol{X})\right)\right)^T = \left(\frac{\partial F_r(\boldsymbol{X})}{\partial r_{0,1}} \ \frac{\partial F_r(\boldsymbol{X})}{\partial r_{0,2}} \ \frac{\partial F_r(\boldsymbol{X})}{\partial r_{0,3}} \right)$$

if the observations only depend on the geocentric position vector and are referring to only one epoch. The gradient only depends on the type of observations used, whereas the second term is independent of the observation type and related to the **variational equations**. This separates the observation-specific (**geometric**) part from the **dynamic** part.



Variational Equations

For each orbit parameter P_i the corresponding **variational equation** reads as

$$\ddot{\boldsymbol{z}}_{P_i} = \boldsymbol{A}_0 \cdot \boldsymbol{z}_{P_i} + \boldsymbol{A}_1 \cdot \dot{\boldsymbol{z}}_{P_i} + \frac{\partial \boldsymbol{f}_1}{\partial P_i}$$

with $\boldsymbol{z}_{P_i}(t) \doteq \frac{\partial \boldsymbol{r}_0}{\partial P_i}(t)$ and the 3 x 3 matrices defined by

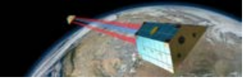
$$A_{0[i;k]} \doteq \frac{\partial f_i}{\partial r_{0,k}} \qquad \text{and} \qquad A_{1[i;k]} \doteq \frac{\partial f_i}{\partial \dot{r}_{0,k}}$$

f_i i -th component of the total acceleration $\boldsymbol{f}$

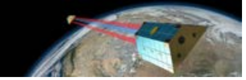
$r_{0,k}$ k-th component of the geocentric position $\boldsymbol{r}_0$

For each orbit parameter P_i the **variational equation** is a linear differential equation system of second order in time. Their solutions are all needed for orbit determination.

(Jäggi, 2007)



GPS-based LEO POD

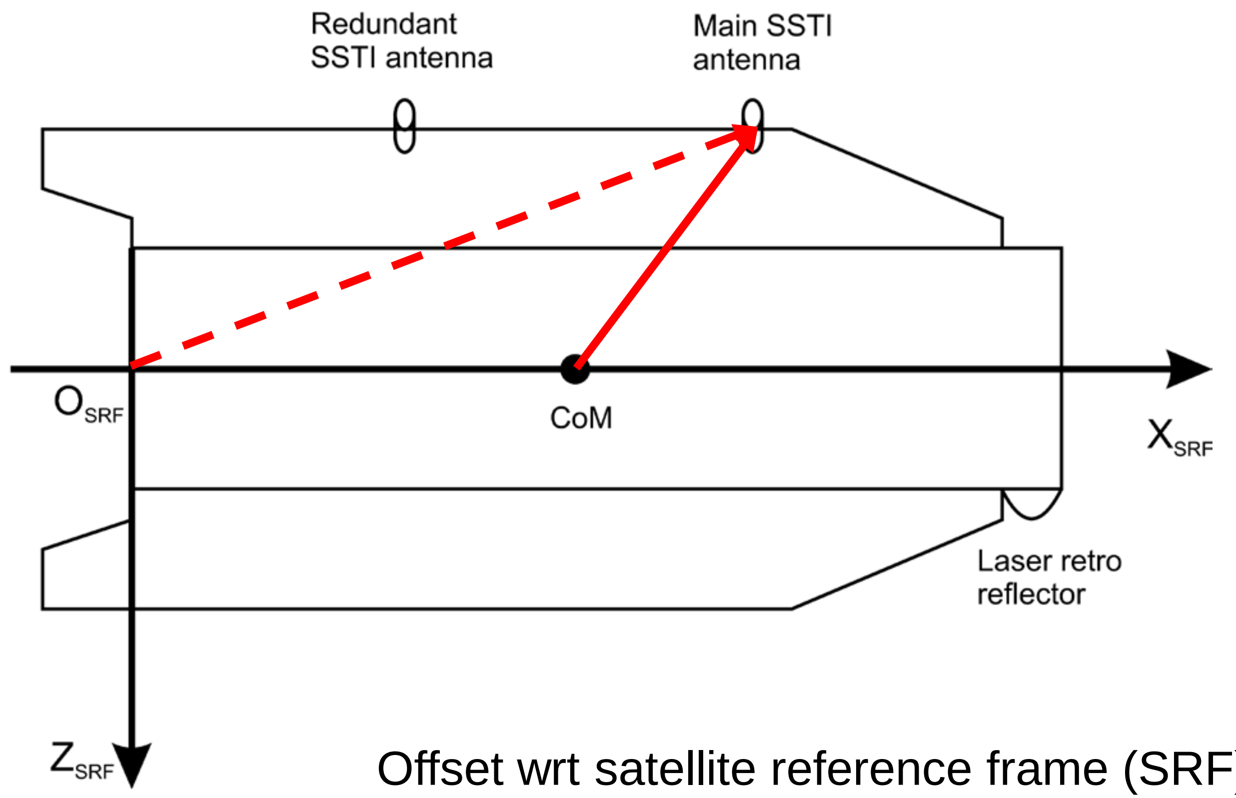


LEO Sensor Offsets

Phase center offsets $\delta\boldsymbol{r}_{leo,ant}$:

- are needed in the inertial or Earth-fixed frame and have to be transformed from the satellite frame using **attitude data** from the star-trackers
- consist of a frequency-independent **instrument offset**, e.g., defined by the center of the instrument‘s mounting plane (CMP) in the satellite frame
- consist of frequency-dependent **phase center offsets** (PCOs), e.g., defined wrt the center of the instrument‘s mounting plane in the antenna frame (ARF)
- consist of frequency-dependent **phase center variations** (PCVs) varying with the direction of the incoming signal, e.g., defined wrt the PCOs in the antenna frame

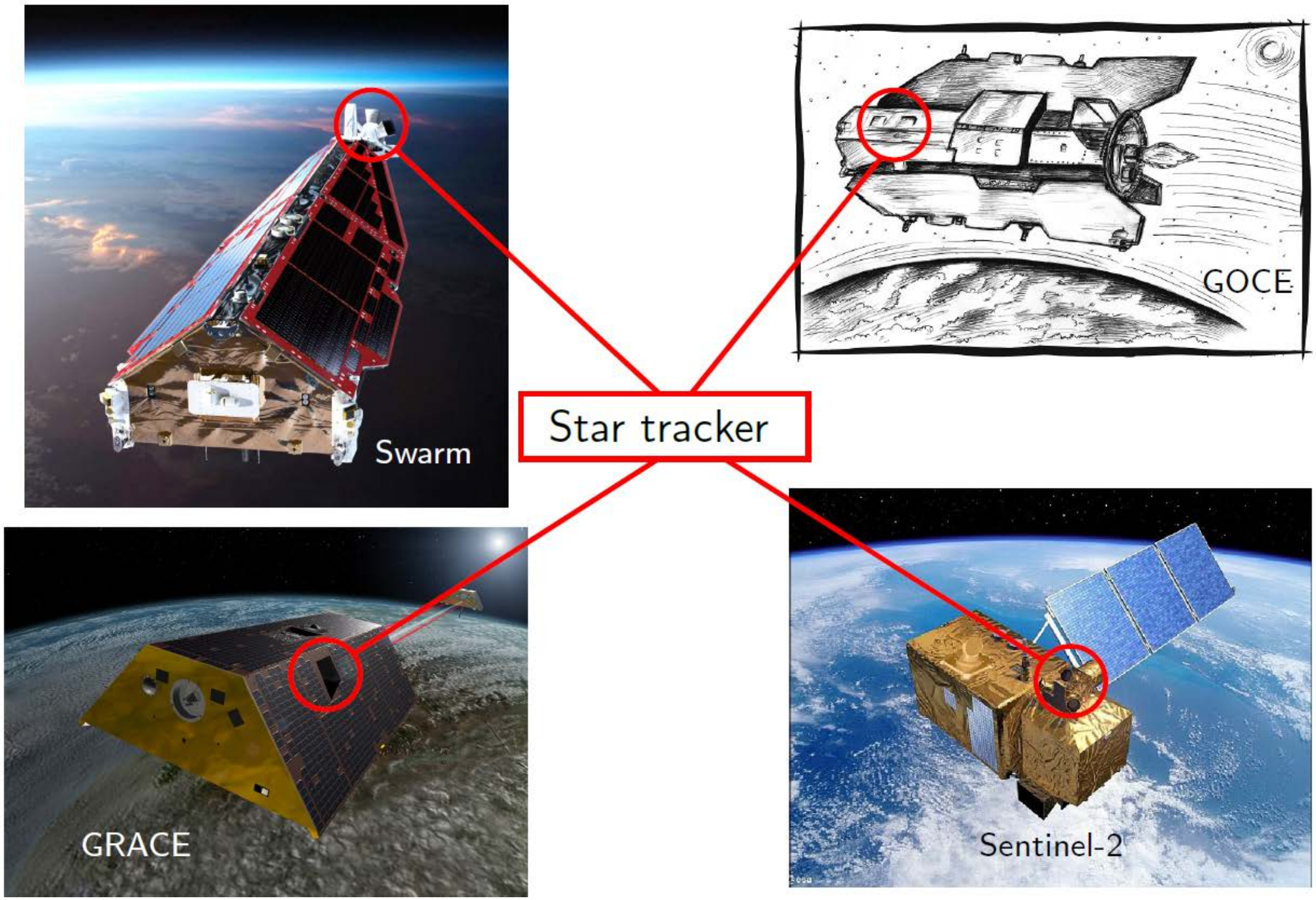
Example: GOCE Sensor Offsets (1)



Offset wrt satellite reference frame (SRF) is **constant**

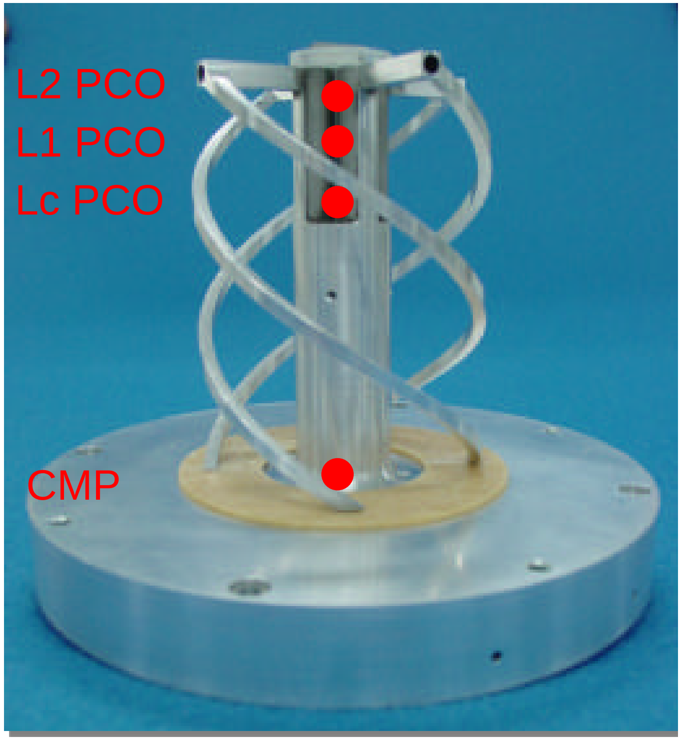
Offset wrt center of mass (CoM) is **slowly varying**

Example: GOCE Sensor Offsets (2)



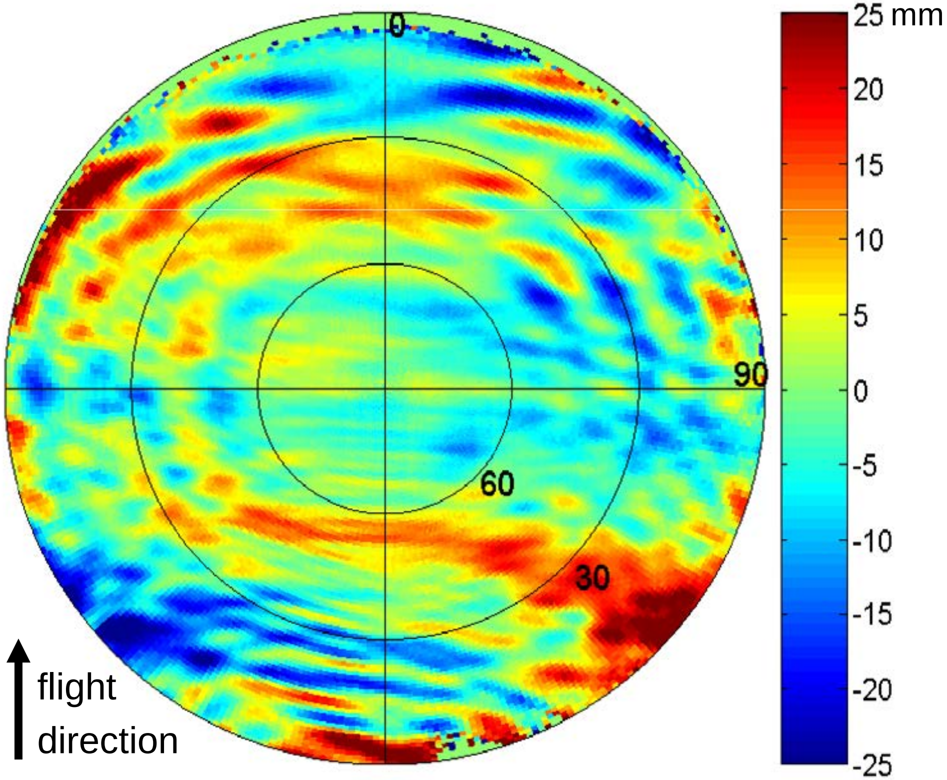
Spaceborne GPS Antennas: GOCE

L1, L2, Lc phase center offsets



Measured from ground calibration in anechoic chamber

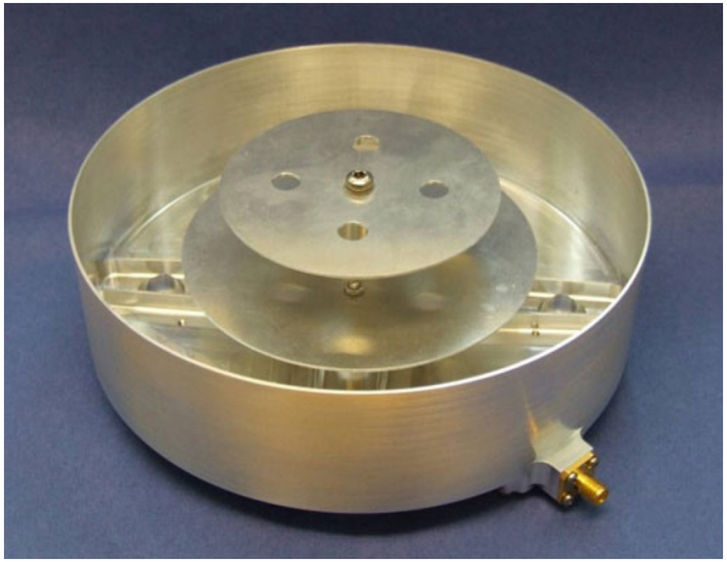
Lc phase center variations



Empirically derived during orbit determination according to Jäggi et al. (2009)

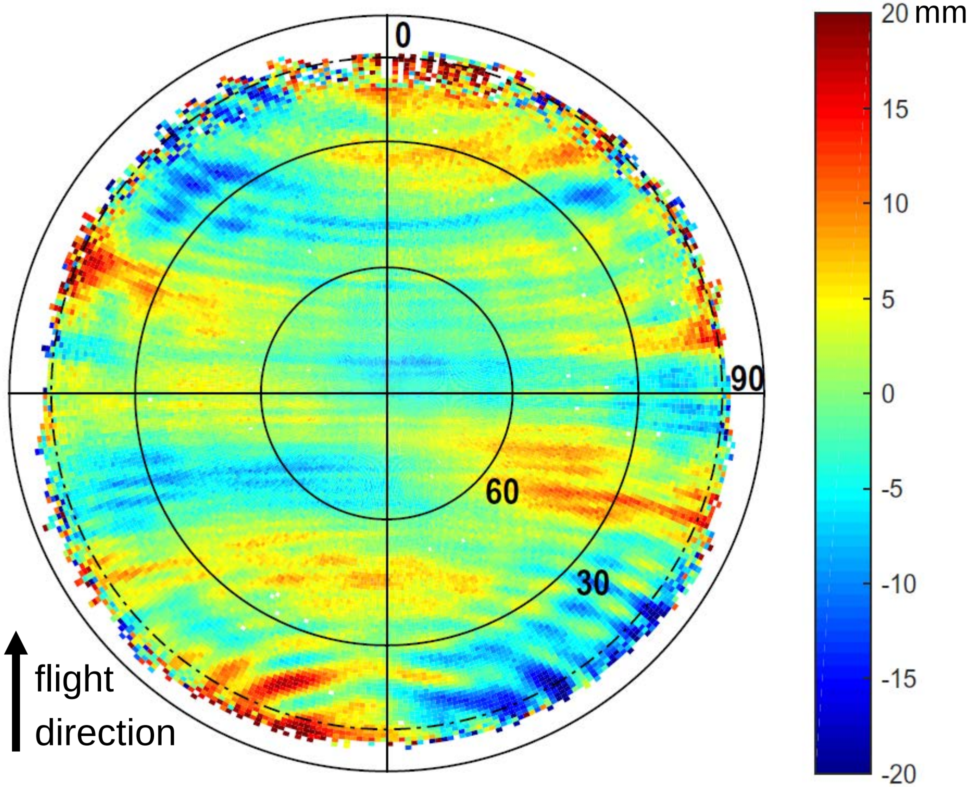
Spaceborne GPS Antennas: Swarm

Swarm GPS antenna



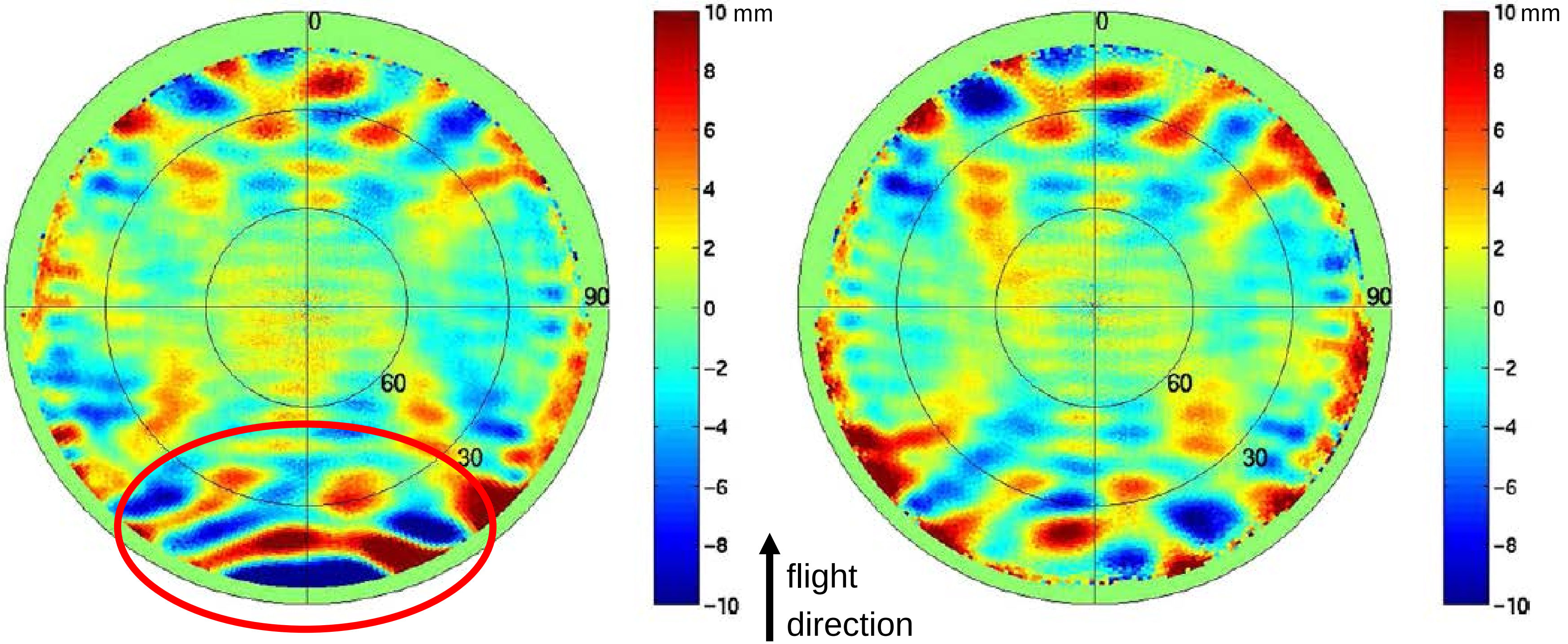
Multipath shall be minimzed by chokering

Lc phase center variations



Empirically derived during orbit determination according to Jäggi et al. (2016)

Spaceborne GPS Antennas: GRACE

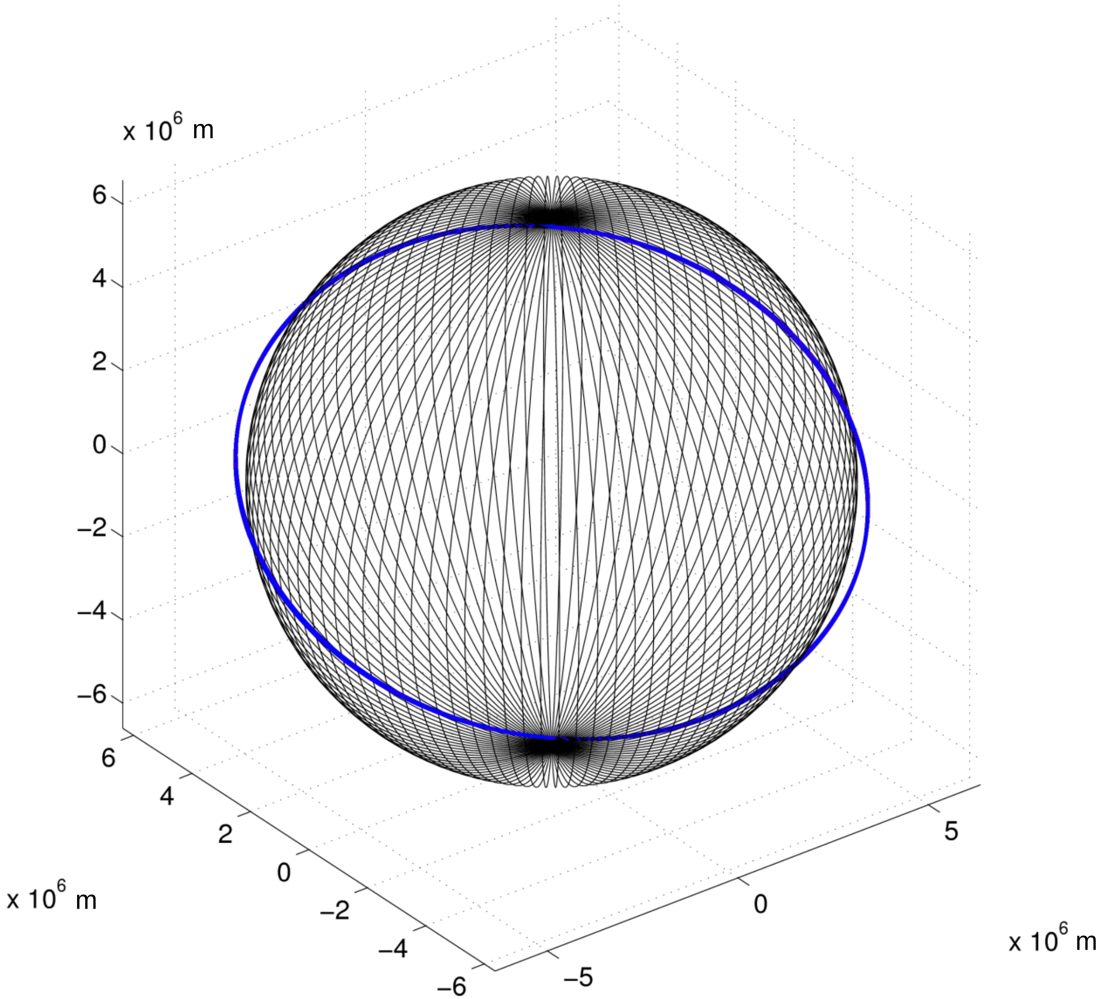


GRACE-A
(occultation antenna switched on)

GRACE-B

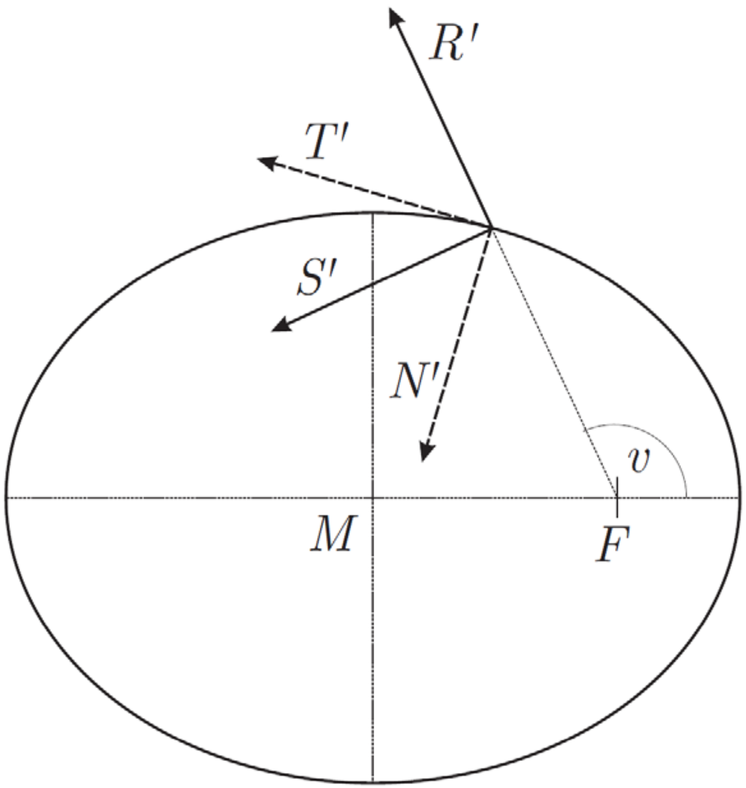
(Jäggi et al., 2009)

Visualization of Orbit Solutions



It is more instructive to look at differences between orbits in well suited coordinate systems …

Co-Rotating Orbital Frames



R, S, C unit vectors are pointing:

- into the radial direction
- normal to **R** in the orbital plane
- normal to the orbital plane (cross-track)

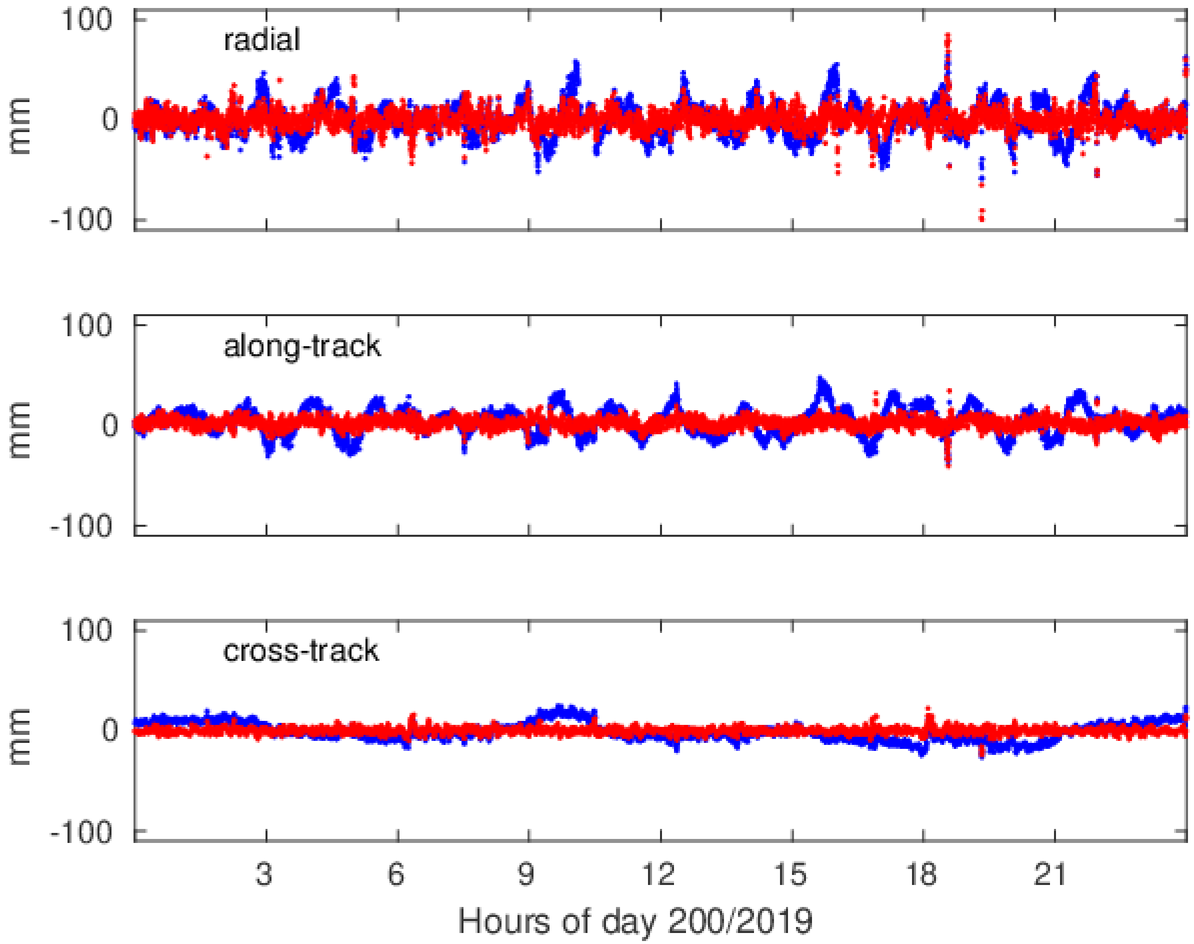
T, N, C unit vectors are pointing:

- into the tangential (along-track) direction
- normal to **T** in the orbital plane
- normal to the orbital plane (cross-track)

Small eccentricities: **S**~**T** (velocity direction)

Orbit Differences KIN-RD (Sentinel-3A)

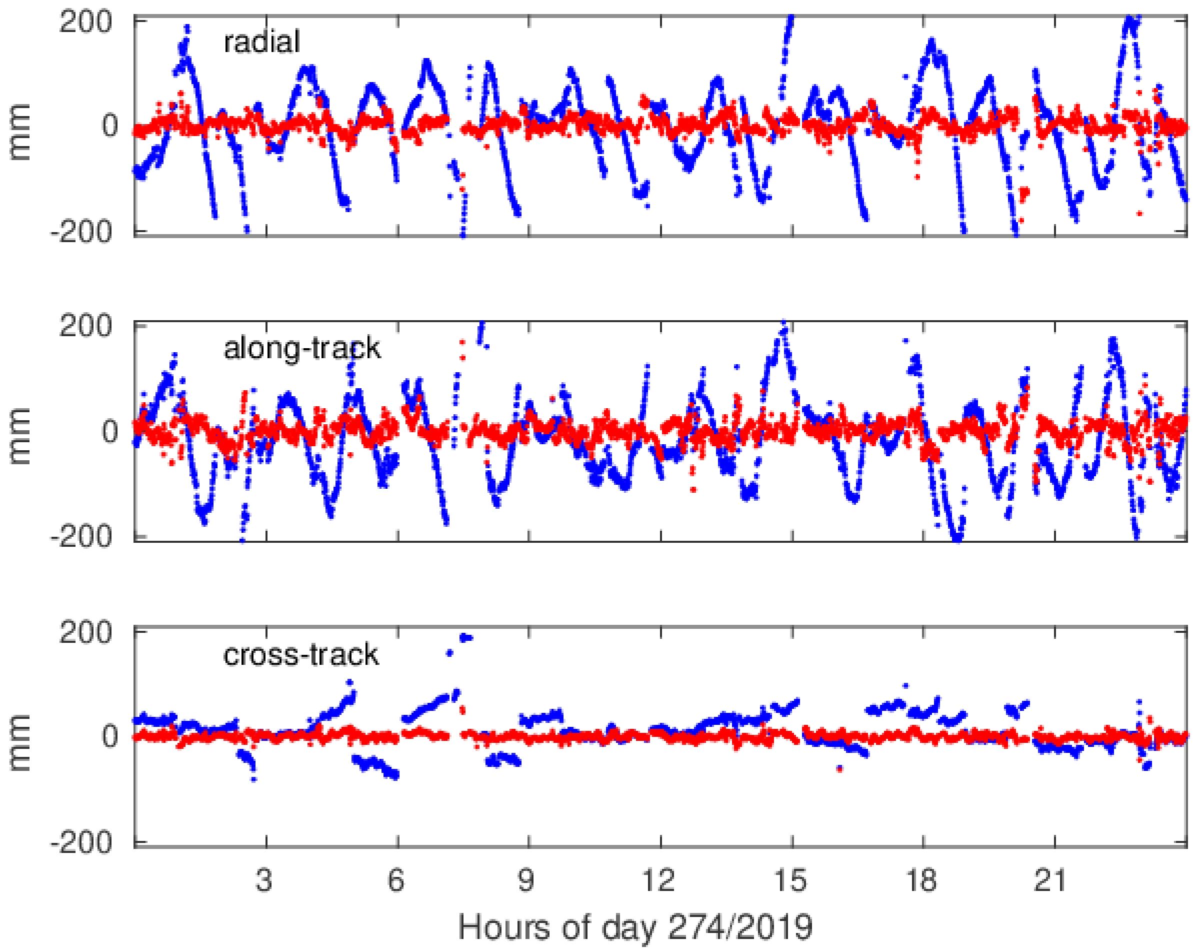
Differences at epochs of kin. positions



Comparison of ambiguity-float solutions and ambiguity-fixed solutions.

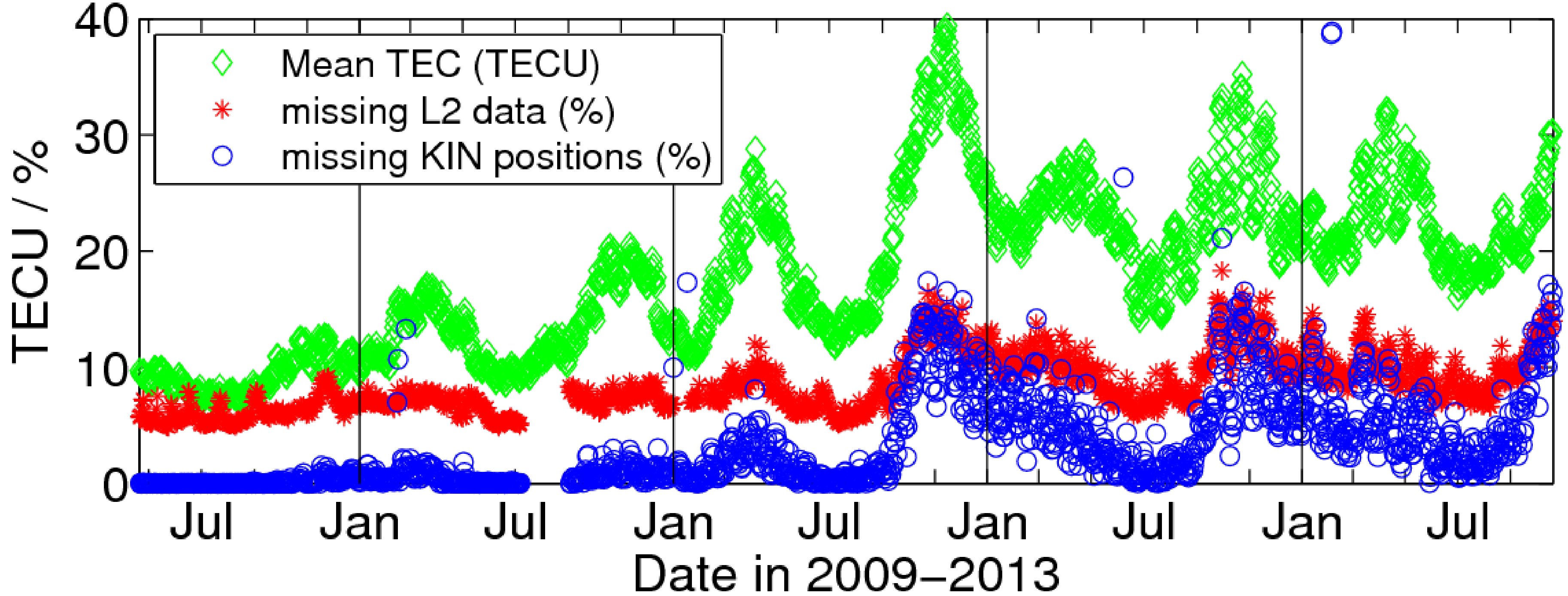
Orbit Differences KIN-RD (COSMIC-2)

Differences at epochs of kin. positions, FM-1, POD-1



Comparison of ambiguity-float solutions and ambiguity-fixed solutions. (Jäggi et al. 2021)

Orbit Differences KIN-RD (GOCE)

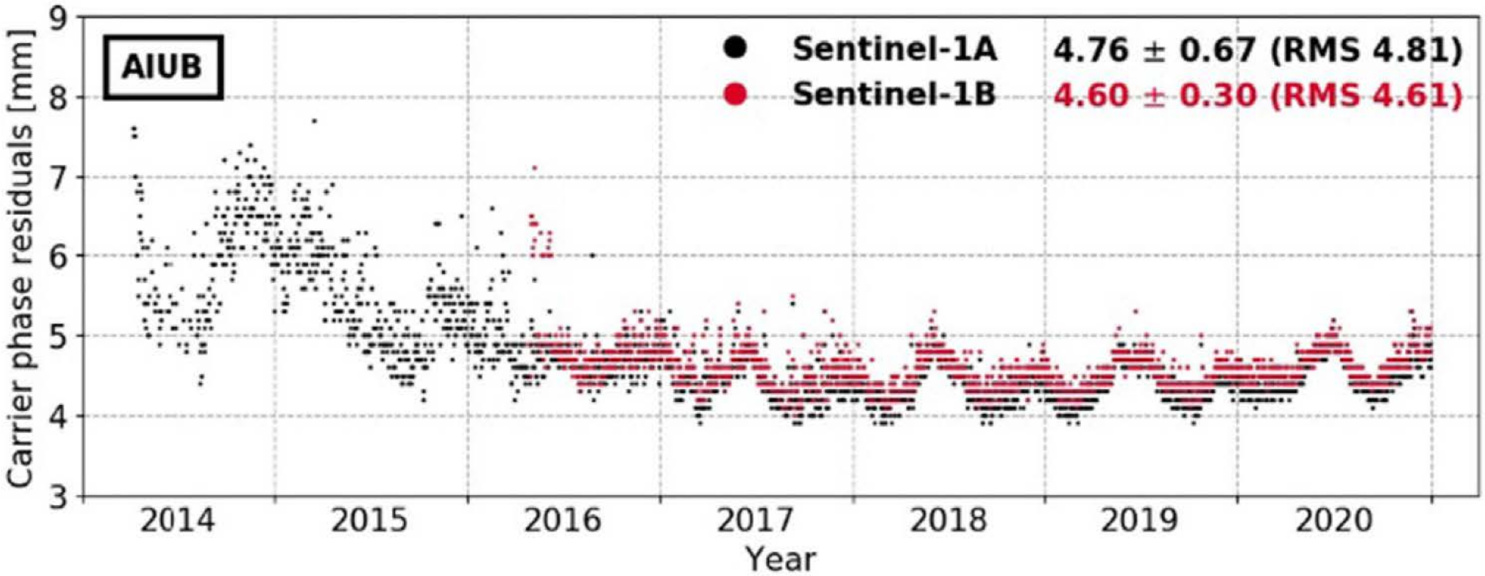


The result illustrates the **consistency** between both orbit-types. The level of the differences is usually given by the quality of the kinematic positions.

The differences are highly correlated with the **ionosphere activity** and with **data losses on L2**.

(Bock et al., 2014)

RMS of Carrier Phase Residuals (Sentinel-1)



The plot illustrates the **quality** of the orbit fit. The level is given by the adopted parametrization, depending on how dynamic the orbit parametrization is.

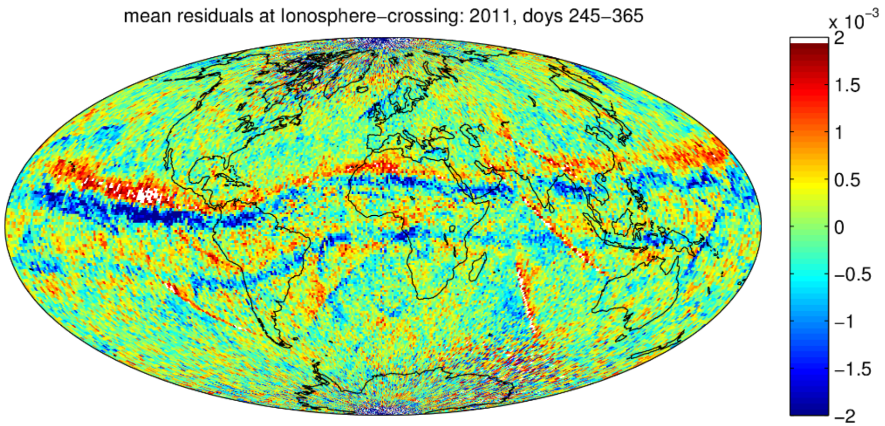
The variations reveal again the impact of the **ionosphere activity** and also further **modeling deficiencies.**

(Fernández et al., 2022)

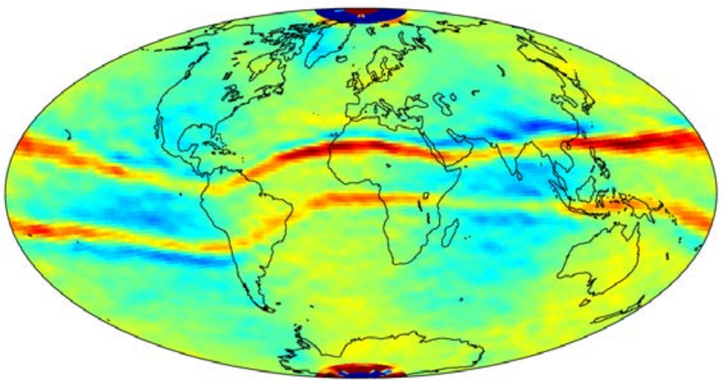
Consequences of Ionospheric Effects in Orbits

For GOCE systematic effects around the geomagnetic equator were observed in the ionosphere-free GPS phase residuals **=> affects kinematic positions**

Degradation of kinematic positions around the geomagnetic equator propagates into gravity field solutions.



Phase observation residuals (- 2 mm … +2 mm) mapped to the ionosphere piercing point

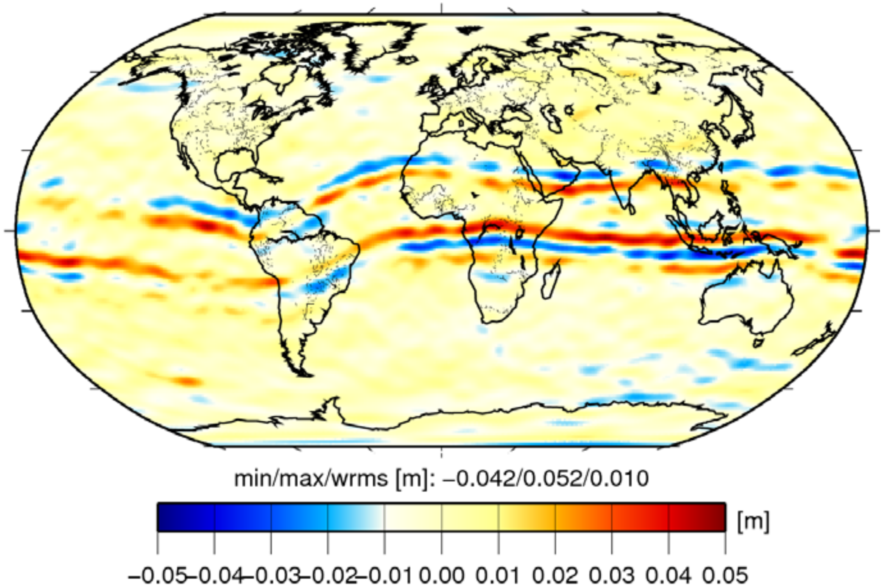


Geoid height differences (−5 cm … 5 cm);
R4 period

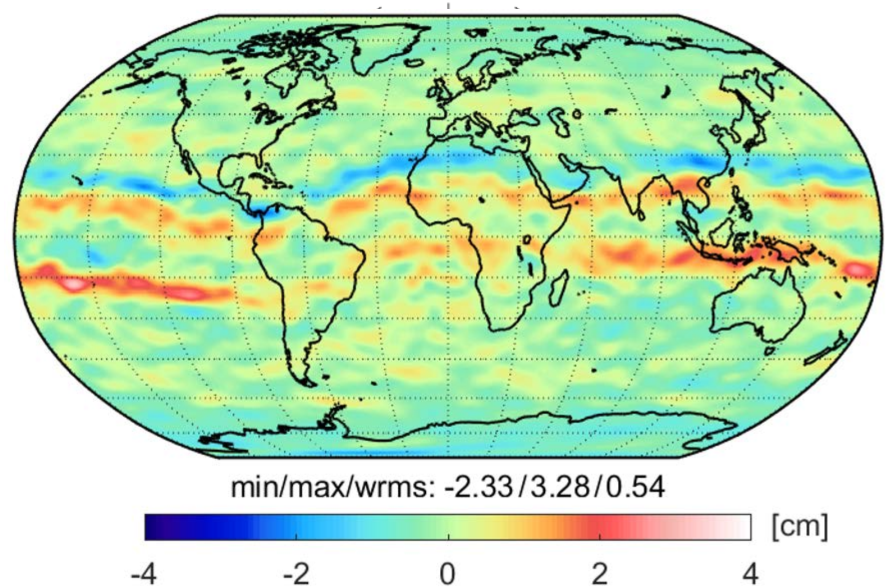
(Jäggi et al., 2015)

Systematic Errors in GPS Data (1)

Original GPS Data (Swarm)



Original GPS Data (GRACE)

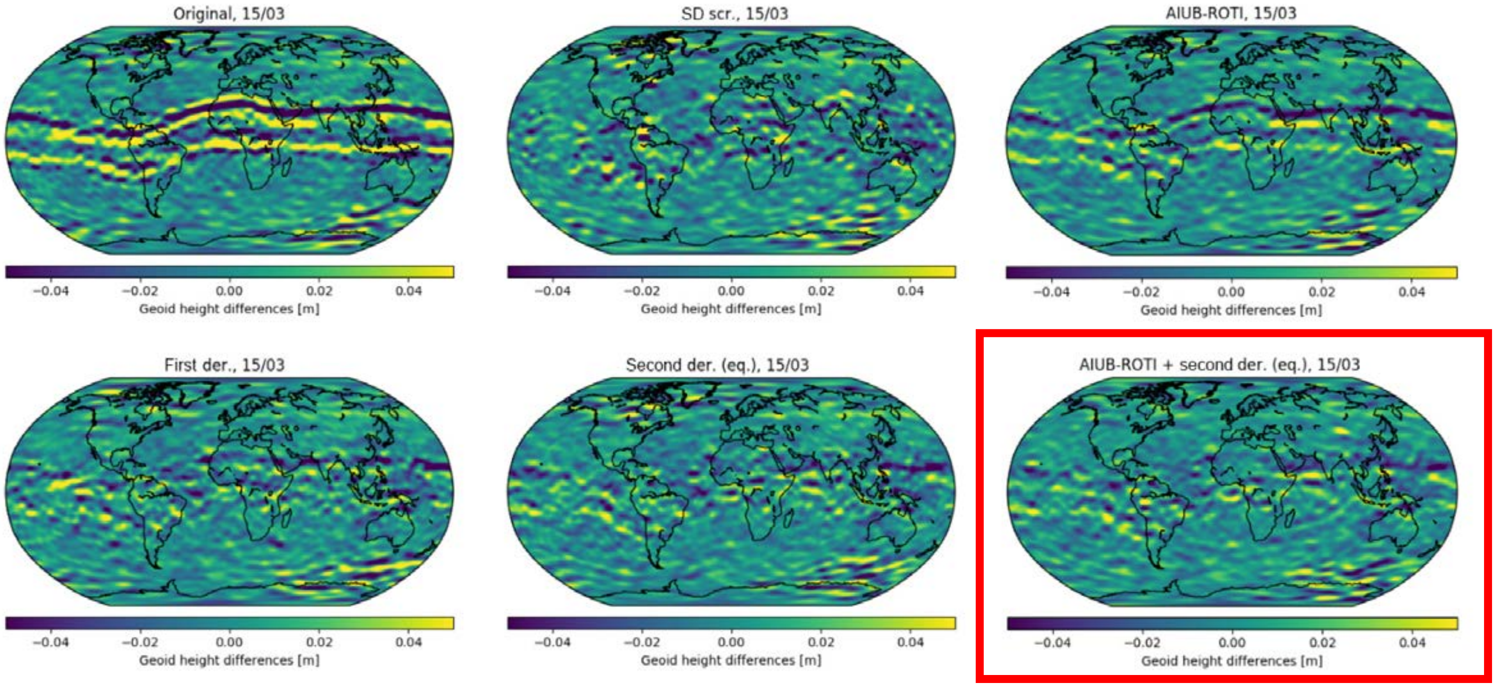


(Differences wrt GOCO05S, 400 km Gauss smoothing adopted)

Systematic signatures along the geomagnetic equator are **"not"** visible when using original L1B RINEX GPS data files from the GRACE mission.

(Jäggi et al., 2016)

Systematic Errors in GPS Data (2)

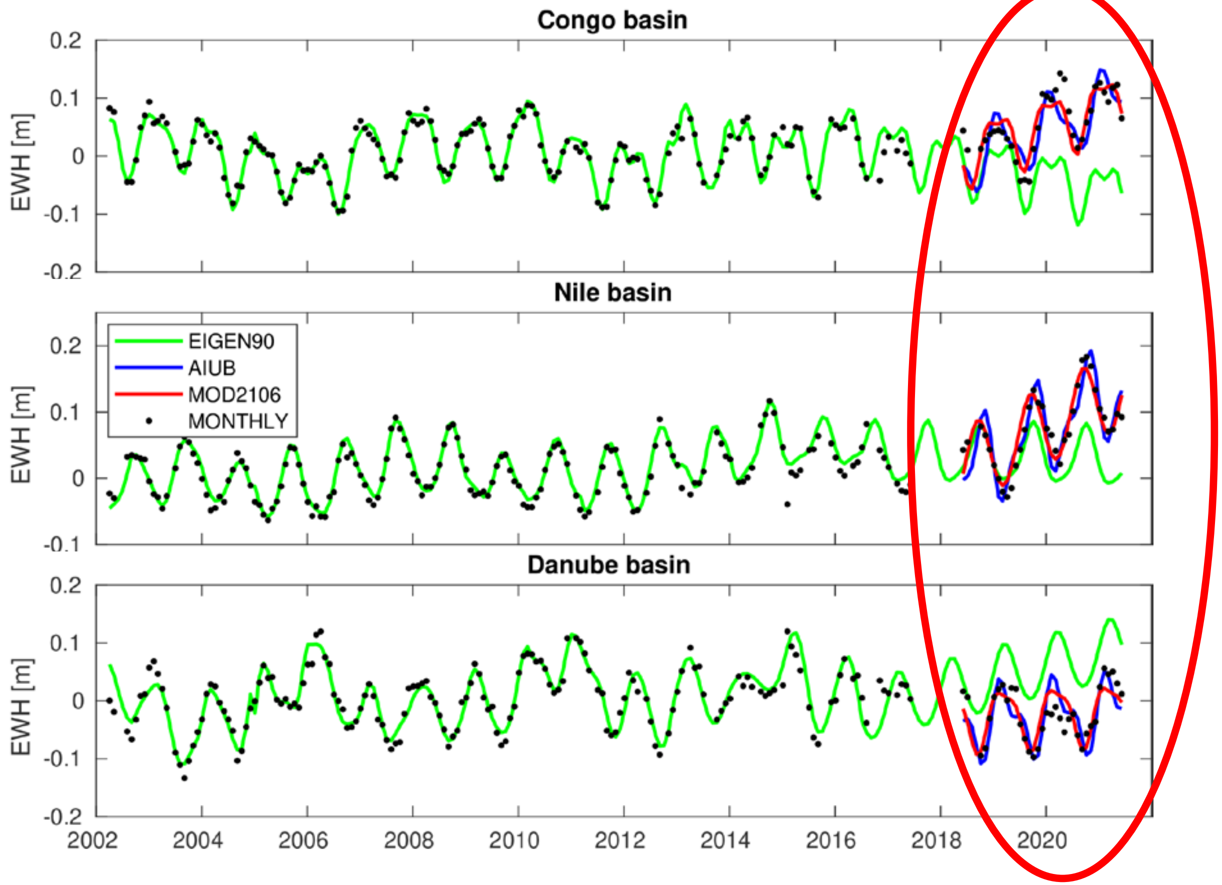


(Differences wrt JPL-GRACE-RL06, 400 km Gauss smoothing adopted)

Systematic signatures along the geomagnetic equator may be efficiently reduced when down-weighting the GPS data using derivatives of the geometry-free linear combination. ROTI-based down-weighting additionally reduces scintillation noise.

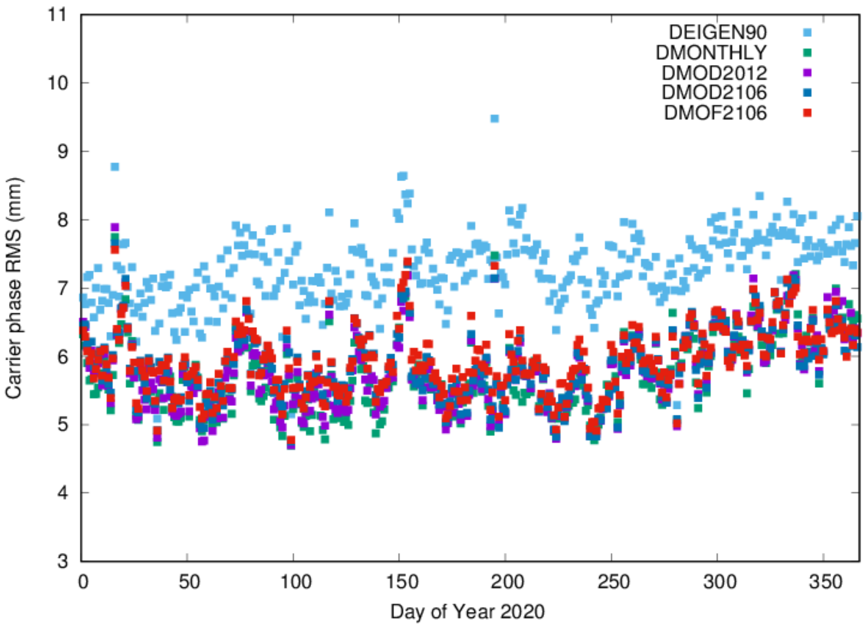
(Schreiter et al., 2019)

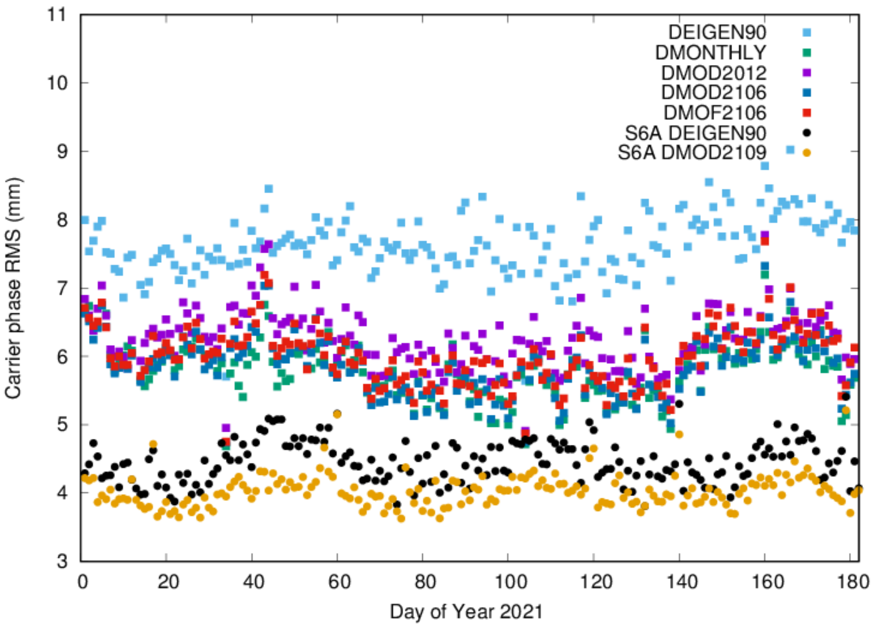
Sensitivity to the Time-Variable Gravity Field Model (1)



The predictions of the EIGEN-GRGS.RL04 model (containing no data after 2017) are rather poor as shown here for three river basins. The new fitted signal model (FSM) to COST-G monthly solutions has clear advantages. (Peter et al., 2022)

Sensitivity to the Time-Variable Gravity Field Model (2)





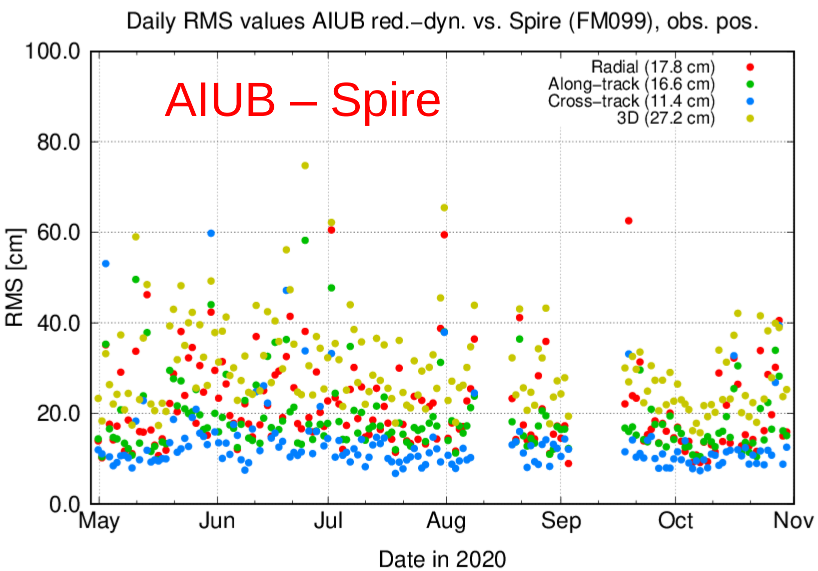
Monthly COST-G GRACE-FO gravity fields (▪) outperform GRACE-based static fields (▪) with co-estimated time-variations for Sentinel-3B (left and right plot).

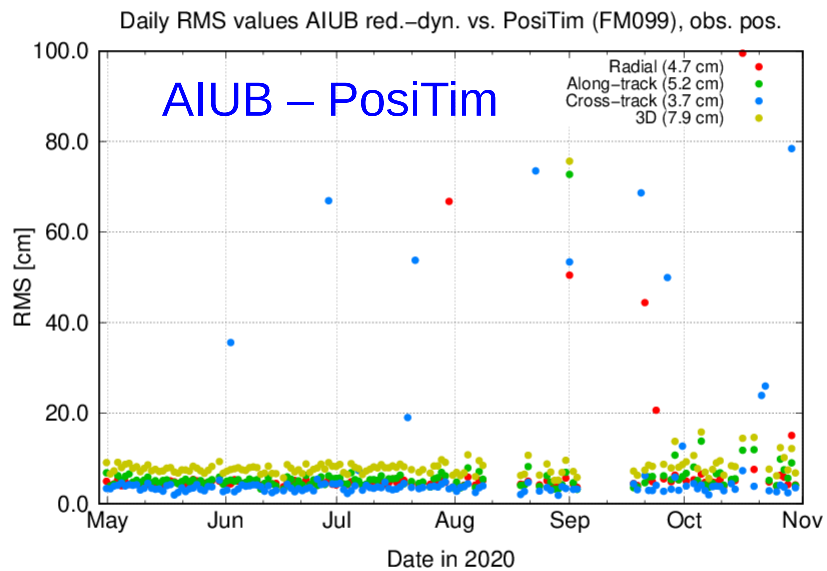
COST-G fitted signal models (FSM) perform comparably good (▪ ▪ ▪, left plot), which also holds for Sentinel-6A (●, right plot). In particular the COST-G FSM show also good prediction capabilities (▪, right plot).

(Peter et al., 2022)

Orbit Validation

Orbit Comparisons (Spire FM099)



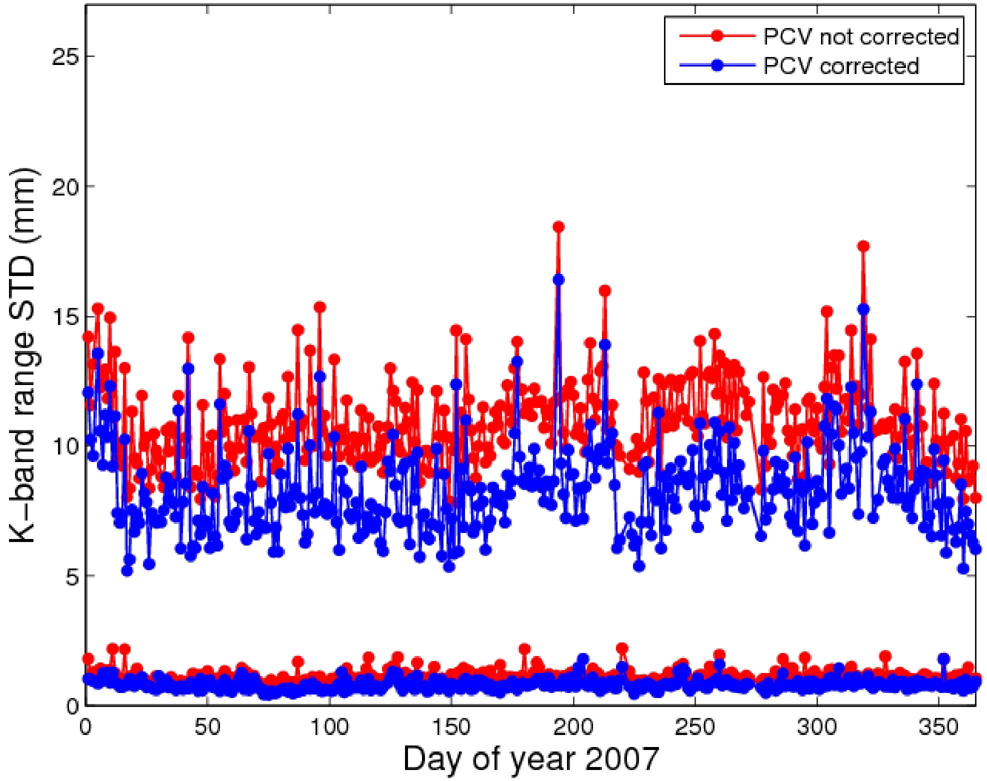


Orbit comparisons between solutions computed **with different software packages** are helpful, especially if no external orbit validations are possible.

The plots show for the example of one Spire satellite that orbit differences AIUB – PosiTim are significantly smaller than the orbit differences AIUB – Spire.

(Jäggi et al., 2022)

GRACE Orbit Validation with K-Band



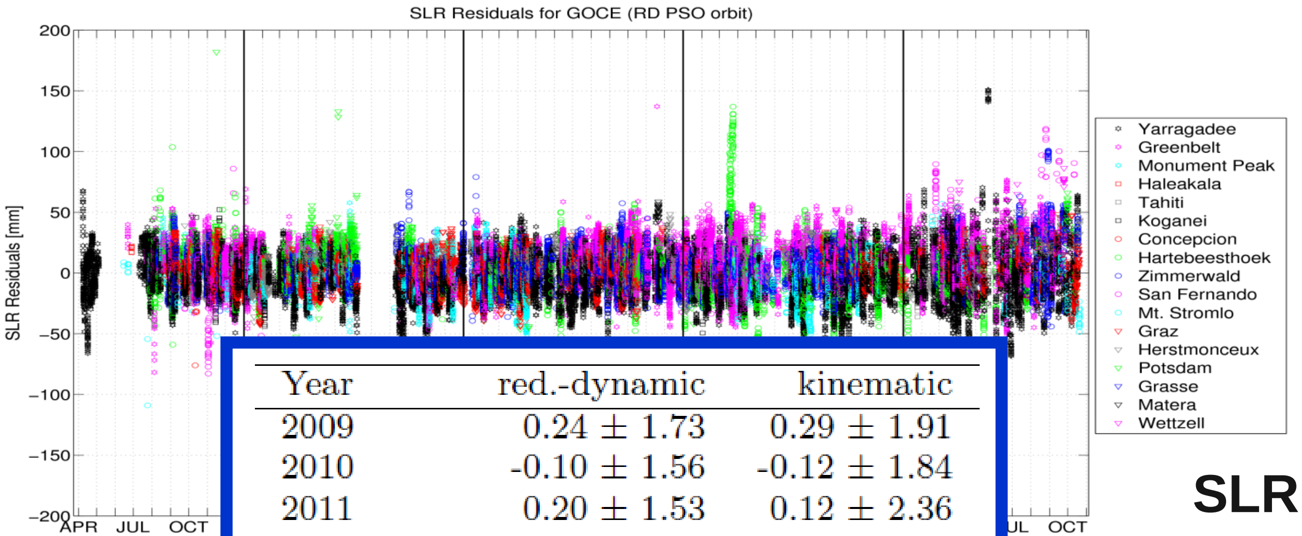
ZD solution
< 1cm

DD solution
(ambiguity-fixed)
< 1mm

The ultra-precise and continuously available K-Band data allow it to validate the **inter-satellite distances** between the GRACE satellites. Thanks to this validation, e.g., PCV maps were recognized to be crucial for high-quality POD.

(Jäggi et al., 2009)

Orbit Validation with SLR (GOCE)

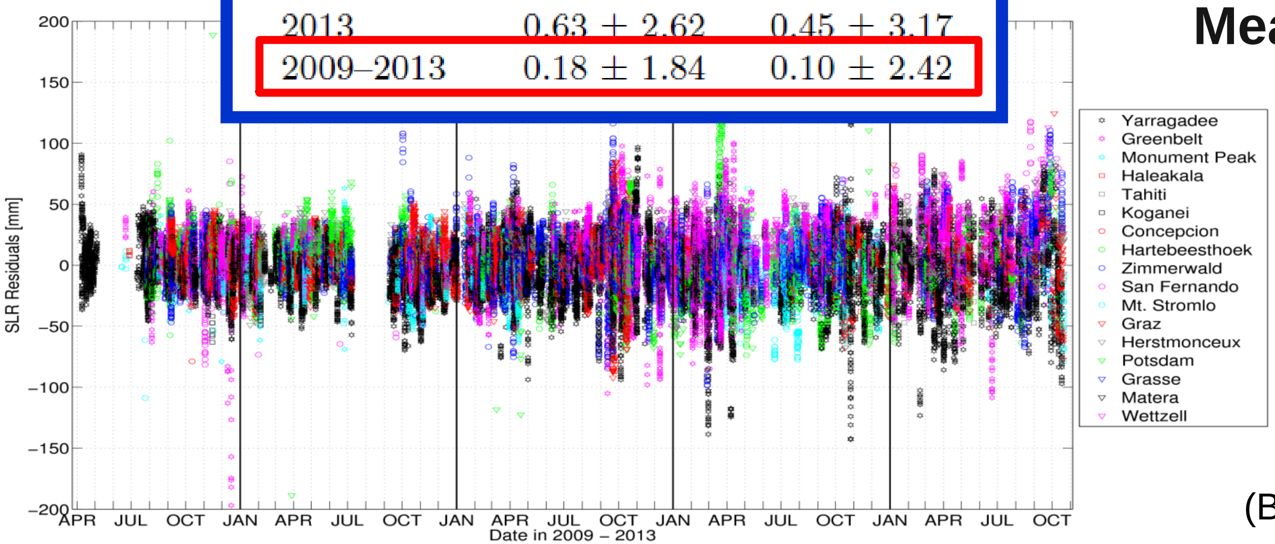


Reduced-dynamic

Year	red.-dynamic	kinematic
2009	0.24 ± 1.73	0.29 ± 1.91
2010	-0.10 ± 1.56	-0.12 ± 1.84
2011	0.20 ± 1.53	0.12 ± 2.36
2012	0.10 ± 1.94	-0.05 ± 2.78
2013	0.63 ± 2.62	0.45 ± 3.17
2009–2013	0.18 ± 1.84	0.10 ± 2.42

SLR statistics:

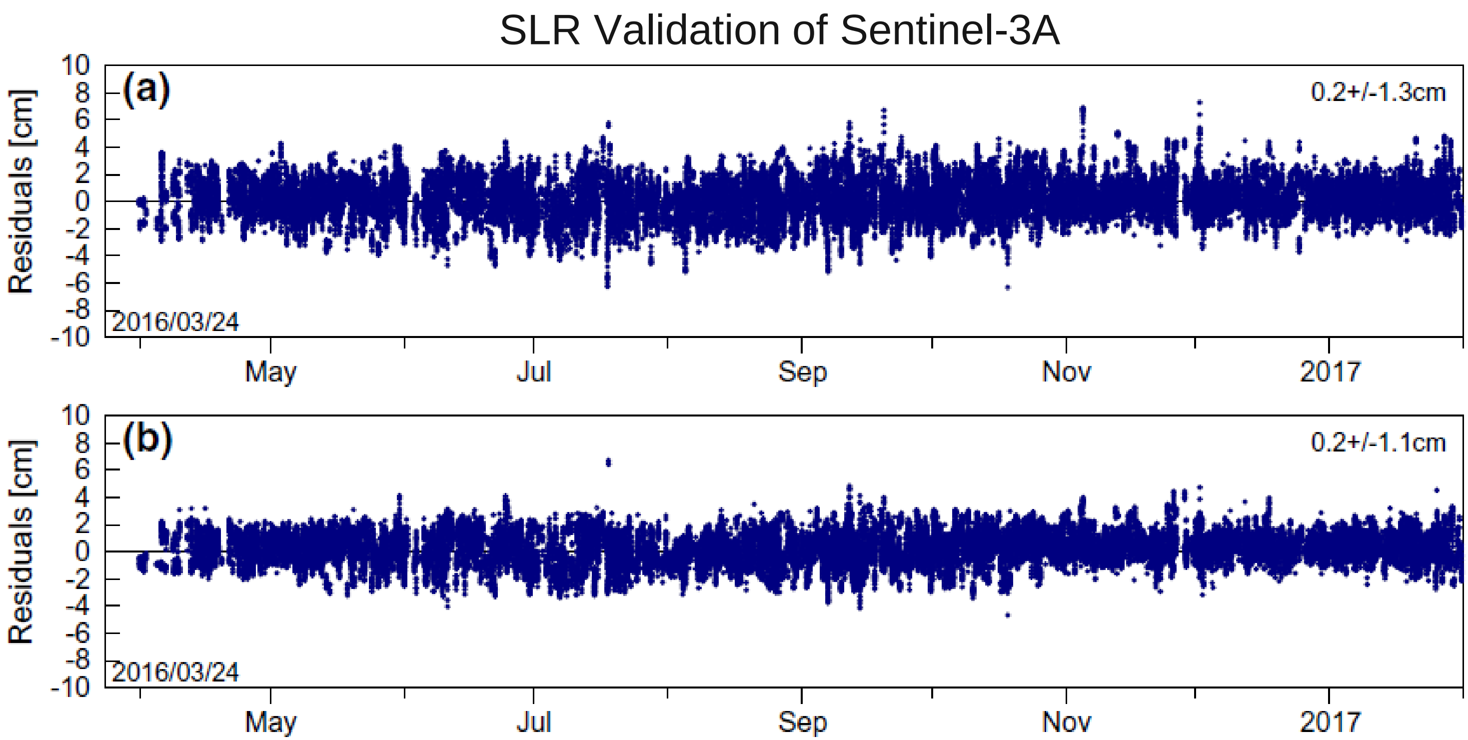
Mean ± RMS (cm)



Kinematic

(Bock et al., 2014)

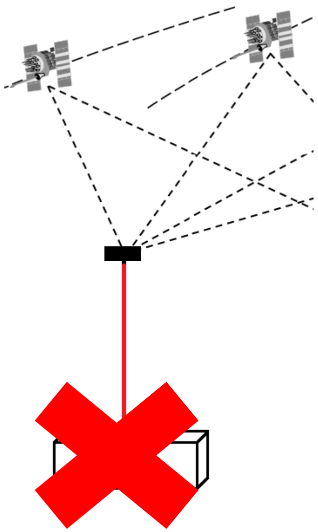
Impact of Undifferenced Ambiguity Resolution (1)

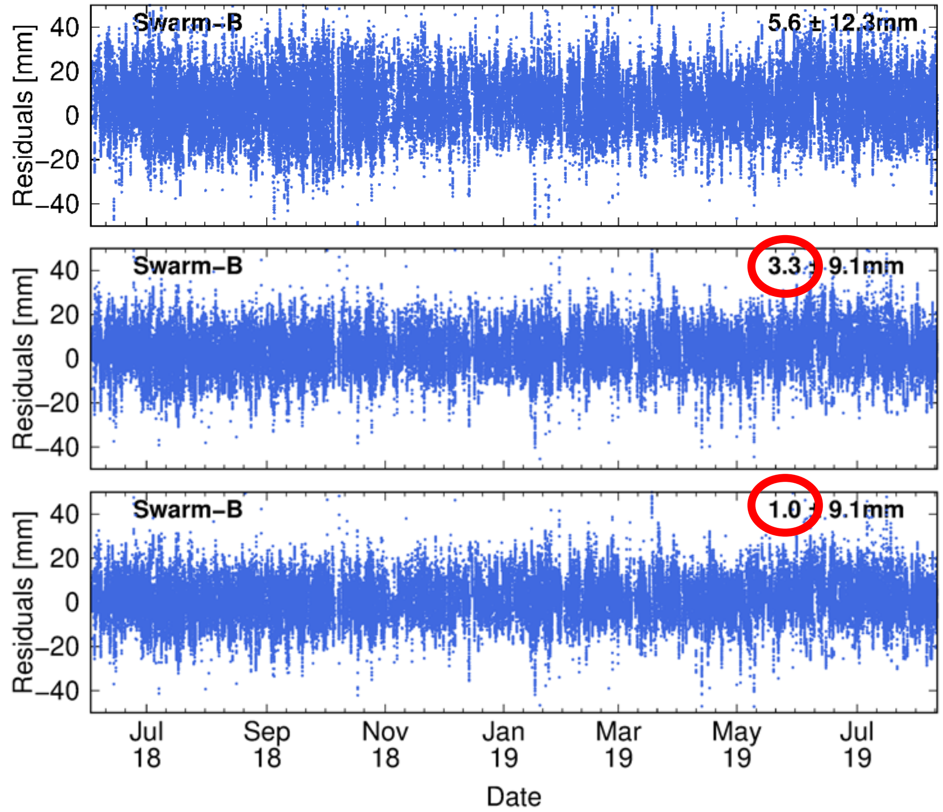


Single-receiver ambiguity fixing may be enabled by using phase bias products and corresponding clock products provided by the IGS analysis centers without the need to form any baselines. It allows to identify lateral offsets in the GPS antenna or center-of-mass location and to significantly stabilize the LEO trajectories.

(Montenbruck et al., 2018)

Impact of Undifferenced Ambiguity Resolution (2)





Ambiguity-float, no non-grav. modeling

Ambiguity-fixed, no non-grav. modeling

Ambiguity-fixed, with non-grav. modeling

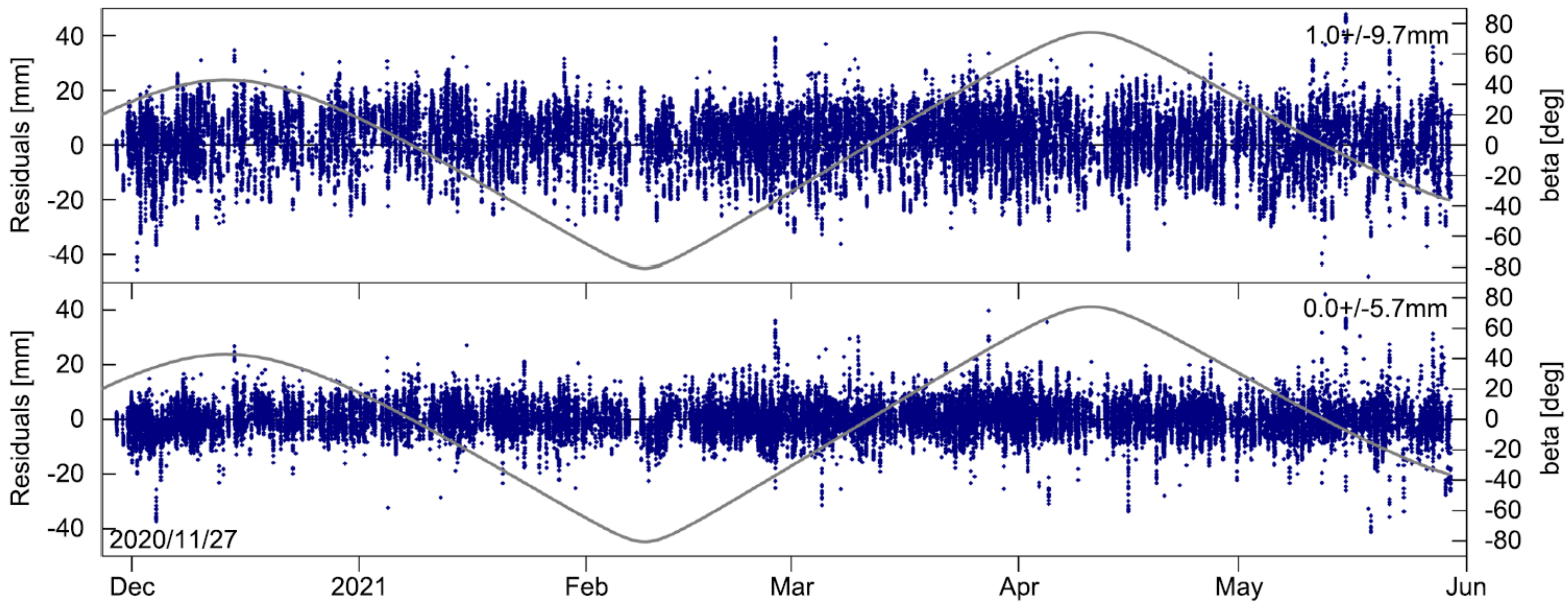
LEO POD significantly profits from single-receiver ambiguity fixing techniques and high-quality signal-specific phase bias products, e.g., by Schaer et al. (2021).

(Arnold et al., 2019; Mao et al., 2021)

Multi-GNSS LEO POD

SLR Validation of Sentinel-6A

Station offsets adjusted



While **Galileo** measurements **exhibit 30–50% smaller RMS errors** than those of GPS, the POD benefits most from the availability of an increased number of satellites. For Sentinel-6A a 1-cm consistency of ambiguity-fixed GPS-only and Galileo-only solutions with the dual-constellation orbits can be demonstrated.

(Montenbruck et al., 2021)

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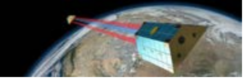

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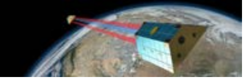

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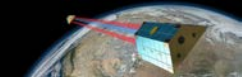

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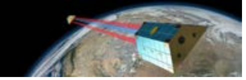

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Pocket Guide of Least-Squares Adjustment (1)

The system of **Observation Equations** is given by:

$$\boldsymbol{L}' + \boldsymbol{\epsilon} = \boldsymbol{F}(\boldsymbol{X})$$

or, if $\boldsymbol{F}$ is a non-linear function of the parameters, in its **linearized** form:

$$\boldsymbol{L}' + \boldsymbol{\epsilon} = \boldsymbol{F}(\boldsymbol{X}_0) + \boldsymbol{A}\,\boldsymbol{x}$$

$\boldsymbol{L}'$	Tracking observations	$\boldsymbol{X}_0$	A priori parameter values
$\boldsymbol{\epsilon}$	Observation corrections	$\boldsymbol{x}$	Parameter corrections
$\boldsymbol{F}$	Functional model	$\boldsymbol{X}$	Improved parameter values, i.e., $\boldsymbol{X} = \boldsymbol{X}_0 + \boldsymbol{x}$

$$\boldsymbol{A} \doteq \left.\frac{\partial \boldsymbol{F}(\boldsymbol{X})}{\partial \boldsymbol{X}}\right|_{\boldsymbol{X}=\boldsymbol{X}_0}$$ First design matrix

Pocket Guide of Least-Squares Adjustment (2)

The system of **Normal Equations** is obtained by minimizing $\boldsymbol{\epsilon}^T \, P \, \boldsymbol{\epsilon}$:

$$\left(\boldsymbol{A}^T \, \boldsymbol{P} \, \boldsymbol{A}\right) \, \boldsymbol{x} - \boldsymbol{A}^T \, \boldsymbol{P} \, \boldsymbol{l} = \boldsymbol{N} \, \boldsymbol{x} - \boldsymbol{b} = \mathbf{0}$$

$\boldsymbol{N} \doteq \boldsymbol{A}^T \, \boldsymbol{P} \, \boldsymbol{A}$	Normal equation matrix
$\boldsymbol{b} \doteq \boldsymbol{A}^T \, \boldsymbol{P} \, \boldsymbol{l}$	Right-hand side with "O-C" term $\boldsymbol{l} \doteq \boldsymbol{L}' - \boldsymbol{F}(\boldsymbol{X}_0)$
$\boldsymbol{P} = \sigma_0^2 \, \boldsymbol{C}_{\boldsymbol{ll}}^{-1}$	Weight matrix, from covariance matrix $\boldsymbol{C}_{\boldsymbol{ll}}$ of observations

For a **regular** normal equation matrix the parameter corrections follow as:

$$\boldsymbol{x} = \left(\boldsymbol{A}^T \, \boldsymbol{P} \, \boldsymbol{A}\right)^{-1} \, \boldsymbol{A}^T \, \boldsymbol{P} \, \boldsymbol{l} = \boldsymbol{N}^{-1} \, \boldsymbol{b}$$

Pocket Guide of Least-Squares Adjustment (3)

The **a posteriori standard deviation of unit weight** is computed as:

$$m_0 = \sqrt{\frac{\boldsymbol{\epsilon}^T \, P \, \boldsymbol{\epsilon}}{f}}$$

f Degree of freedom (number of observations minus number of parameters)

The **covariance matrix** of the adjusted parameters is given by

$$\boldsymbol{C_{xx}} = m_0^2 \, \boldsymbol{Q_{xx}} = m_0^2 \, \boldsymbol{N}^{-1}$$

and their a posteriori standard deviations follow from the diagonal elements:

$$m_x = \sqrt{C_{xx}} = m_0 \, \sqrt{Q_{xx}}$$

Pocket Guide of Least-Squares Adjustment (4)

Parameter pre-elimination is useful to handle a large number of parameters efficiently. Let us sub-divide the system of normal equations into two parts:

$$\begin{pmatrix} \boldsymbol{N}_{11} & \boldsymbol{N}_{12} \\ \boldsymbol{N}_{21} & \boldsymbol{N}_{22} \end{pmatrix} \cdot \begin{pmatrix} \boldsymbol{x}_1 \\ \boldsymbol{x}_2 \end{pmatrix} = \begin{pmatrix} \boldsymbol{b}_1 \\ \boldsymbol{b}_2 \end{pmatrix}$$

We we may reduce the normal equation system by pre-eliminating epoch-specific parameters $\boldsymbol{x}_2$, which yields the modified system of normal equations as

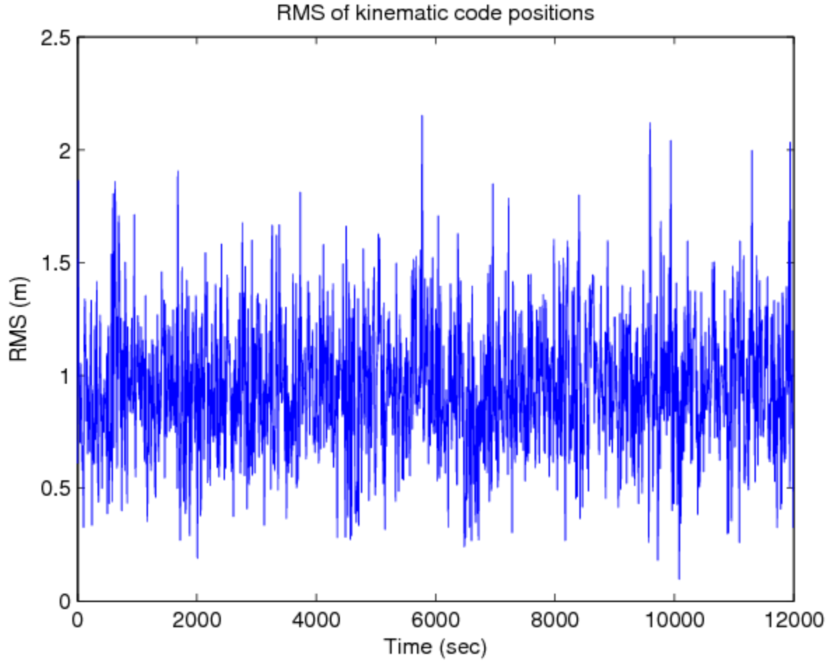
$$\boldsymbol{N}_{11}^{*}\,\boldsymbol{x}_1 = \boldsymbol{b}_1^{*}$$

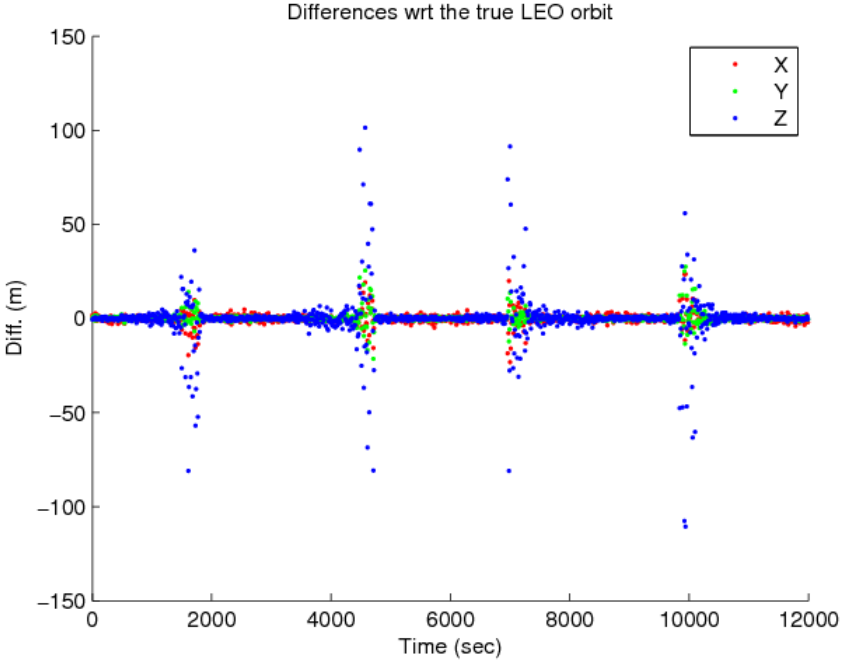
where

$$\boldsymbol{N}_{11}^{*} = \boldsymbol{N}_{11} - \boldsymbol{N}_{12}\,\boldsymbol{N}_{22}^{-1}\,\boldsymbol{N}_{21}$$ is the normal equation matrix of $\boldsymbol{x}_1$

$$\boldsymbol{b}_1^{*} = \boldsymbol{b}_1 - \boldsymbol{N}_{12}\,\boldsymbol{N}_{22}^{-1}\,\boldsymbol{b}_2$$ is the corresponding right-hand side of the normal equation system

Main Results from Lab 2 (1)

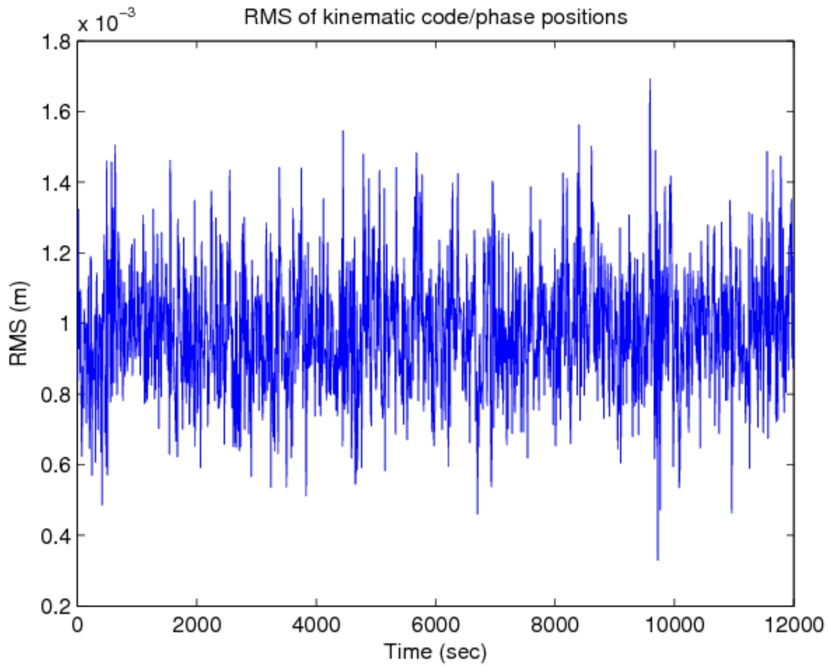


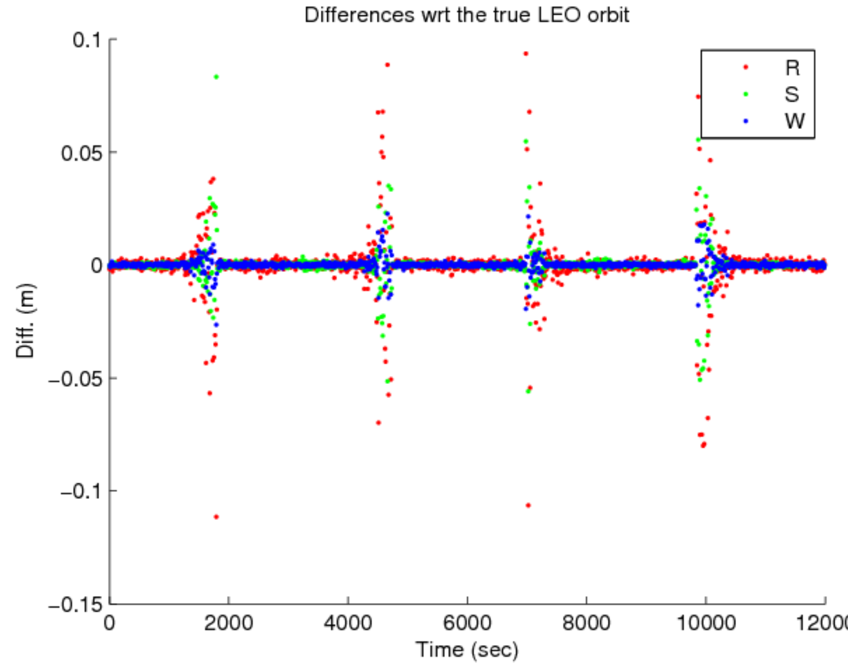


The **a posteriori RMS of unit weight** (m_0) of a Code-only kinematic LEO point positioning varies between about 0.5m and 2m (left).

The differences between the estimated kinematic positions and the true LEO orbit show large deviations twice-per-revolution (right).

Main Results from Lab 2 (2)





The **a posteriori RMS of unit weight** (m_0) of a combined Code- and Phase kinematic LEO point positioning varies between about 0.6 and 1.4mm (left).

Due to the phase observations the quality of the kinematic positions has reached **sub-cm accuracies** for most epochs. Problematic phases twice-per-revolution, however, are still visible.